SUBJECT:

Required maintenance for the Battery Relocation (P/N 350-700324 and 355-700324)

APPLICABILITY:

Aircraft with the subject modification embodied in accordance with TCCA STC No. SH96-31 or any relevant foreign approvals.

The information and data contained in this document supersede or supplement that contained in the basic AS 350 /AS 355 Maintenance documentation in those areas listed herein. For procedures not contained in this document refer to the Approved Maintenance Manual or any other accepted supplemental Maintenance Manual Supplemental.

This MMS is to be used in conjunction with the Approved AS 350 / AS 355 Maintenance Manual for the aircraft with the subject design change incorporated.

The information and data contained in this document supersede or supplement that contained in the basic AS 350 / AS 355 Maintenance documentation in those areas listed herein. For procedures not contained in this document refer to the Approved Maintenance Manual or any other Supplemental Instructions for Continued Airworthiness.

This Supplemental ICA is to be used in conjunction with the Approved AS 350 / AS 355 Maintenance Manual for the aircraft with the subject design change incorporated.

The Airworthiness Limitations section is FAA approved and specifies maintenance required under 14 CFR Secs. 43.16 and 91.403 unless an alternative program has been FAA approved.

	NAME AND SIGNATURE	DATE	COMPANY DEPARTMENT
PREPARED BY:	Deborah Kerr	Digitally signed by Deborah Kerr DN: cn=Deborah Kerr, o=Airbus, ou, email=deborah.kerr@airbus.com, c=CA Date: 2024.07.19 09:43:31 -04'00'	AHCA ENGINEERING
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CHECKED BY:	Derek Riemer	Digitally signed by Derek Riemer Date: 2024.07.19 14:33:28 -04'00'	AHCA QUALITY ASSURANCE
REV. 13 RELEASED BY:	Jeremy Le Roux	Digitally signed by Jeremy Le Roux Date: 2024.07.19 19:52:37 -04'00'	AHCA ENGINEERING
REV. 13 ACCEPTED (Civil A/W Authority)	(As per ICA Compliance Check Sheet)		TCCA

RECORD OF REVISIONS

Rev.	Pages at this Revision	Description, Reason Changed Pages	Prepared (name and date)	Checked (name and date)	App'd/Acc'd (Civil A/W Authority) (name and date)	Released (name and date)
0	1 through 15	Original issue.	D. Kerr 29 July 2004	C. Timmins 29 July 2004	N/A	R. Manson 4 August 2004
1	1 through 15	Changes to pages 1 to 15. Revised General, Inspection Schedule and all Figure titles as per TCCA request	D. Kerr 8 September 2004	C. Timmins 8 September 2004	TCCA E. Cheung 8 September 2004	R. Manson 8 September 2004
2	1 through 23	Format revised. More detail added to Sections 1, 3, 4, 6, 7, 8 and Weight and Balance. (Pages 4, 5, and 7 to 23)	D. Kerr 27 June 2005	C. Timmins 27 June 2005	N/A	R. Manson 19 July 2005
3	1 through 23	Weight and Balance chart corrected, access door composite layup (Figure 6) added, Control and Operation and placard maintenance clarified, Figure numbers after Figure 6 changed. (Pages 7, 12, 15, and 19 to 24)	D. Kerr 19 July 2005	C. Timmins 19 July 2005	TCCA E. Cheung 19 July 2004	R. Manson 19 July 2005
4	1 through 24 A1 to A10 B1 to B61 C1 to C49	Updated references to Appendix B to reflect document change. (Pages 7, 11, 12, B1 to B61)	D. Kerr 9 September 2007	C. Timmins 13 September 2007	TCCA E. Cheung 22 September 2005	R. Manson 27 September 2005
5	1 through 26 A1 to A10 B1 to B61 C1 to C49	Template updated, SAFT Battery maintenance schedule included. Pages 3 to 6, 9, 10, 12 to 16, 19, 20, 22 to 26 and C1 to C49	D. Kerr 14 June 2007	C. Timmins 14 June 2007	N/A	R. Manson 23 August 2007
6	1 through 27 A1 to A10 B1 to B61 C1 to C49	Put in complete Nickel-Cadmium Aircraft Batteries Operating and Maintenance Manual into Appendix C.	D. Kerr 6 August 2007	C. Timmins 7 August 2007	TCCA F. Eads 23 August 2007	R. Manson 23 August 2007
7	1 through 34	Relocation of relay 35P, and Fuse Holder Assembly. Addition of circuit breaker 102P at STN A1825 and of terminal block plate (TB. P) at tail boom disconnect to coincide with basic a/c configuration. Placards and markings revised. (Pages 4 to 6, 9 to 11, 13, 17, 18, 21 to 25, 27 to 32)	D. Kerr 26 February 2008	C. Timmins 26 February 2008	TCCA F. Eaves 27 February 2008	R. Manson 27 February 2008
8	1 through 34 A1 to A11	Appendix A revised to Revision J. Record of Revisions updated. (Page 4 and Appendix A)	D. Kerr 7 July 2008	C. Timmins 7 July 2008	TCCA F. Eaves 15 July 2008	R. Manson 20 July 2008

NOTE: Revisions to this document will be distributed to operators of this equipment by the STC holder.

NOTE: Revised portions of affected pages are identified by a vertical black line in the margin adjacent to the change.

NOTE: Minor changes are released in accordance with TCCA - ACCEPTED CAR 521.154 procedures (ref. DAPM-E-0001).

RECORD OF REVISIONS

Rev.	Pages at this Revision	Description, Reason Changed Pages	Prepared (name and date)	Checked (name and date)	App'd/Acc'd (Civil A/W Authority) (name and date)	Released (name and date)
9	1 through 44 A1 to A11 B1 to B61	Incorporated AS 355 information into document. New tail boom ballast limits provided and corresponding placard. Wiring diagram revised for a/c with spotlight. Section 4, Inspection Schedule and Maintenance Action revised, 500 flight hours to 600 flight hours. New carbon fiber door available for tail boom. Weight and Balance chart revised. Removed Appendix C. (Pages 4 to 16 18 to 30, 32 to 44)	D. Kerr 13 August 2010	C. Timmins 13 August 2010	TCCA D. Philips 28 October 2010	R. Manson 1 November 2010
10	1 through 53 A1 to A11 B1 to B61	Provide reference to SB 53.00.43 for MOD OP-4309. Wiring drawings revised. Revised Appendix A as per Concorde Component Maintenance Manual Revision L. Section 8 revised. Tail boom ballast label revised. (Pages 4 to 6, 8, 10 to 20, 23 to 29, 31 to 36, 38 to 41 & 43 to 53, all Appendix A)	D. Kerr 7 November 2012	C. Timmins 7 November 2012	TCCA Alex Pompei 19 November 2012	R. Manson 23 November 2012
11	1 through 52 A1 to A33 B1 to B61	Revised the Airworthiness Limitations statement in Section 2. Corrections to Weight and Balance Chart. Labels revised to remove STC references. Rivet callout corrected attaching outer skin doubler. (Pages 5, 6, 7, 11, 20, 23, 41, 43, 46, 47, 51)	D. Kerr 3 May 2013	C. Timmins 3 May 2013	TCCA G. David 8 May 2013	C. Timmins 10 May 2013
12	1 through 24 A1 to A33 B1 to B61	Addition of EASA Airworthiness Limitations statement in Section 2. (Page 23)	D. Kerr 10 September 2013	C. Timmins 11 September 2013	TCCA G. David 24 October 2013	P. Sharpe 24 October 2013
13	1 through 55	Template revised. Compatibility with basic aircraft Mod 07-4836, adding a battery fuse (F1) for variants -01 (PRE MOD 07-4836). For variants -02 (POST MOD 07-4836) a battery Fuse (400A). Updated Wiring Diagrams to include AS 350 (Step 3). Weight and Balance chart revised. Appendix A and B removed from document. (Pages 6 to 15, 17 to 41, 43 to 50, 52)	See page 1.	See page 1.	See page 1.	See page 1.

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NOTE: Minor changes are released in accordance with TCCA - ACCEPTED CAR 521.154 procedures (ref. DAPM-E-0001).

CONTENTS

FIGURES

TABLES

1 GENERAL

A. The existing nickel cadmium battery (15 Ah) is removed from the RH cargo compartment and a high-capacity nickel cadmium battery (22 Ah) or a lead-acid battery (28 Ah) is installed in the tail boom. This eliminates or reduces the need for tail boom ballast and increases the usable volume in the RH cargo compartment. The battery is mounted on a removable tray and is accessible through a cutout approximately 420 mm long x 300 mm wide (16.5 in. long x 11.8 in. wide) in the LH side of the tail boom skin between STN's A 1578 and A 2295. The cutout is locally reinforced by the addition of externally mounted sheet metal doublers. The battery can be accessed through a carbon fiber door attached to the tail boom with two hinges and secured with two latches. Refer to Figures 1 and 2.

The nickel cadmium, Saft 2376 battery type comes with a temperature sensor. The lead-acid, Concorde battery, part number RG-390E is a sealed, valve regulated battery. Refer to Figure 5.

The Battery Relocation consists of the following main components:

Detachable Provisions

- Battery Tray Assembly (Refer to Figure 2)
- Battery (Refer to Figures 5 and 6)
- Access Door (Refer to Figures 14 and 15)

Fixed Provisions

- Battery Harness (Refer to Figure 2)
- Skin Doublers (Refer to Figure 2)
- LH and RH Frame (Refer to Figures 3 and 6)
- Base (Refer to Figures 8 to 13)
- RH and LH Stiffener (Refer to Figures 8 to 13)

For AS 350 PRE MOD 07-3273 and 07-3274, the fuse holder assembly is located in the tail boom between STN's A868 and A1578. Refer to Figure 8.

For AS 350 POST MOD 07-3273 and 07-3274, the fuse holder assembly is located on the battery tray. Refer to Figure 9.

For AS 350 B2/B3 POST MOD 07-4280, addition of Direct Battery Bus Circuit Breaker Box on the original Battery Support Rack area. Refer to Figure 10.

PRE MOD 07-4836, -01 Variant, Relay 35P has been moved forward of the Main Junction Box and the wire harness at the battery has been changed. Refer to Figure 11.

POST MOD 07-4836, -02 Variant has replaced Fuse (F1) with a Terminal Support and Fuse (400A). Refer to Figure 12.

For AS 355 the fuse holder assembly is located in the tail boom between STN's A868 and A1578. Refer to Figure 13.

For AS 350 Harness routing in the tail boom, PRE MOD 07-3273 and 07-3274, refer to Figure 8.

For AS 350 Harness routing in the tail boom, POST MOD 07-3273 and 07-3274, refer to Figure 9.

For AS 350 Harness routing in the tail boom, POST MOD 07-4280 (AS 350 B3), refer to Figure 10.

For AS 350 Harness routing in the tail boom, PRE MOD 07-4836, -01 Variant, refer to Figure 11.

For AS 350 Harness routing in the tail boom, POST MOD 07-4836, -02 Variant, refer to Figure 12.

For AS 355 Harness routing refer to Figure 13.

For instructions for initial installation, see IP-ECL-6 for AS 355 or H240I0221580 for AS 350.

B. These Instructions for Continued Airworthiness are applicable to aircraft with the subject modification embodied

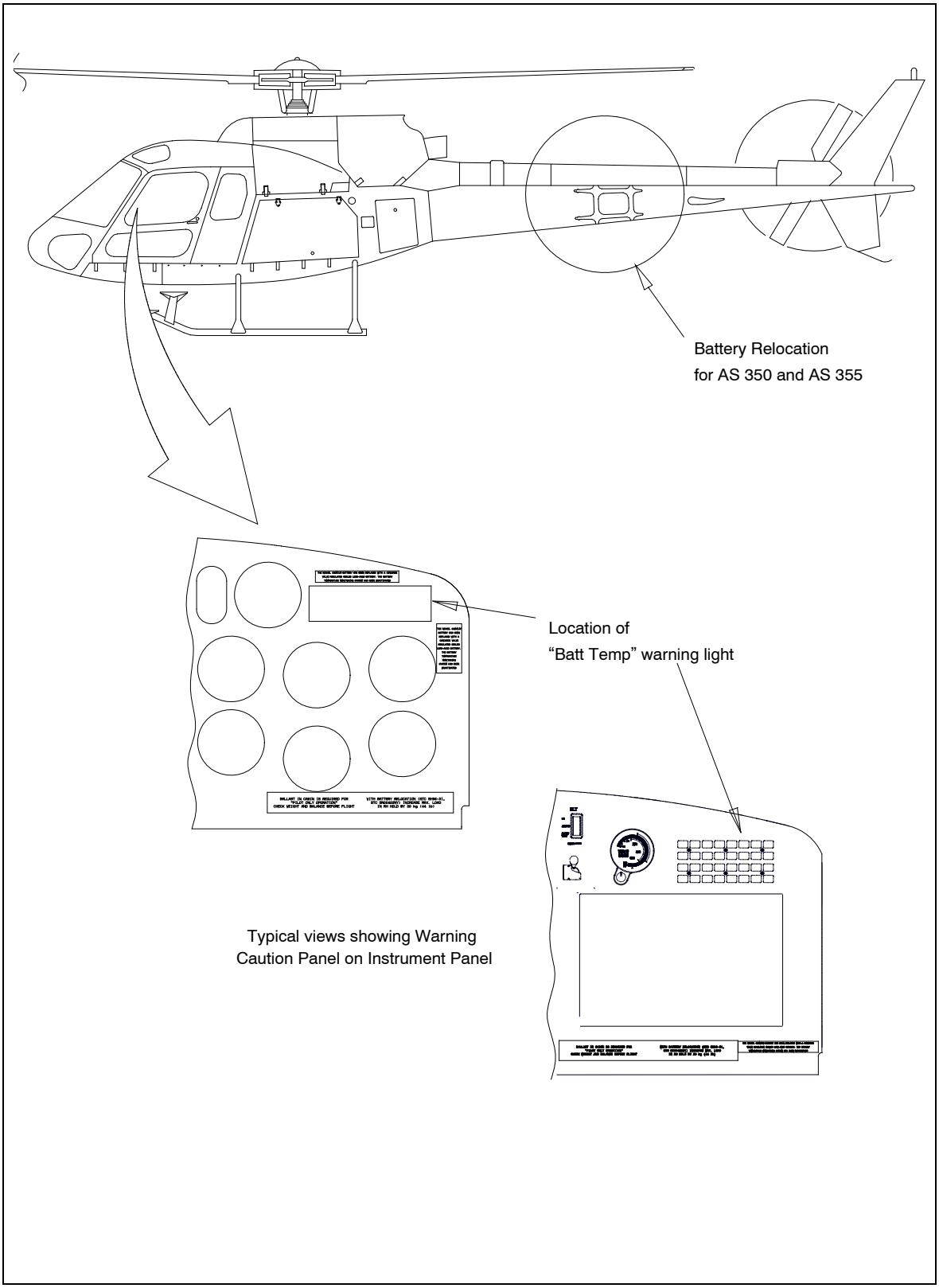


Figure 1 General Layout

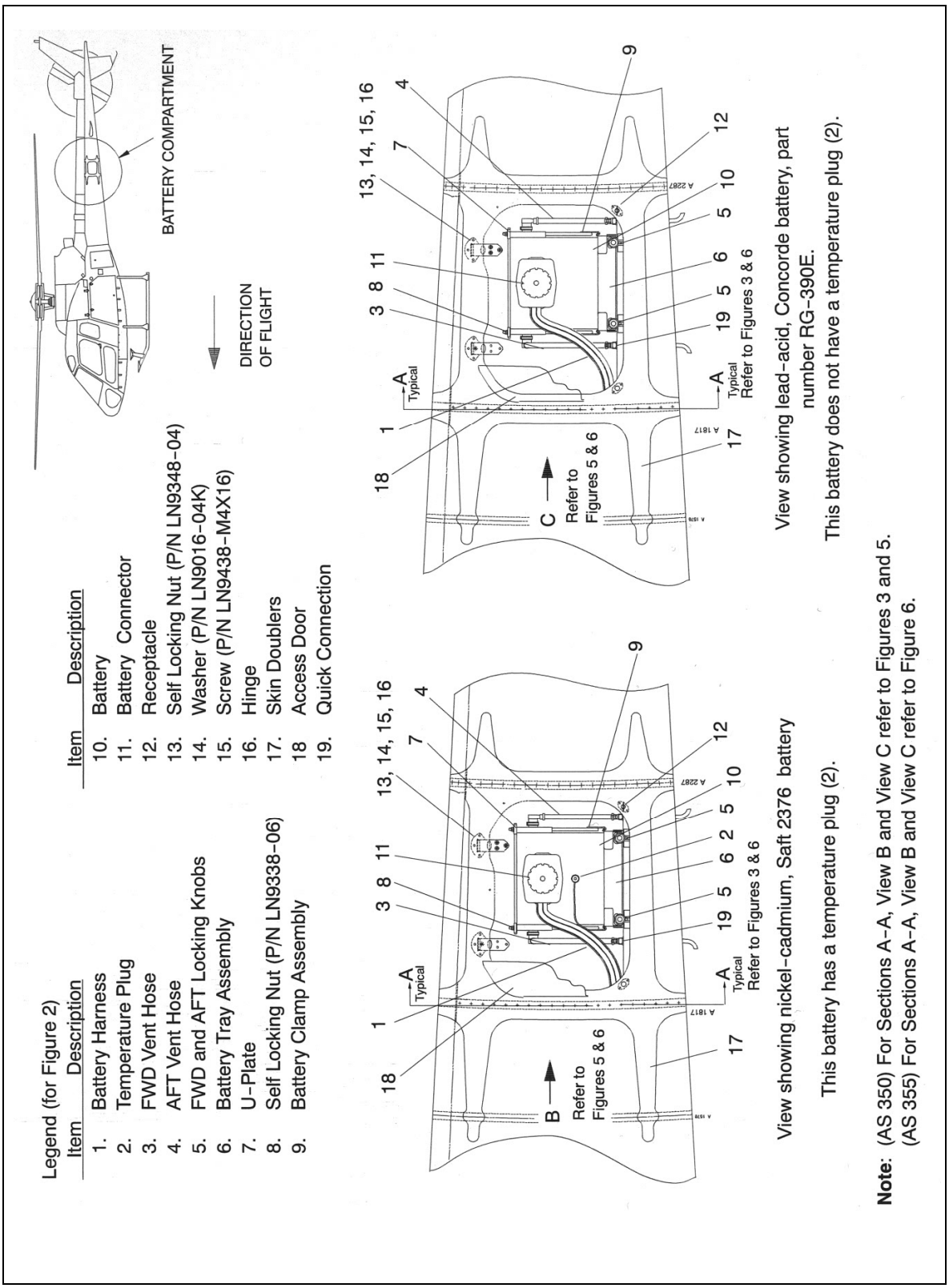


Figure 2 AS 350 and AS 355 Battery Compartment (main views)

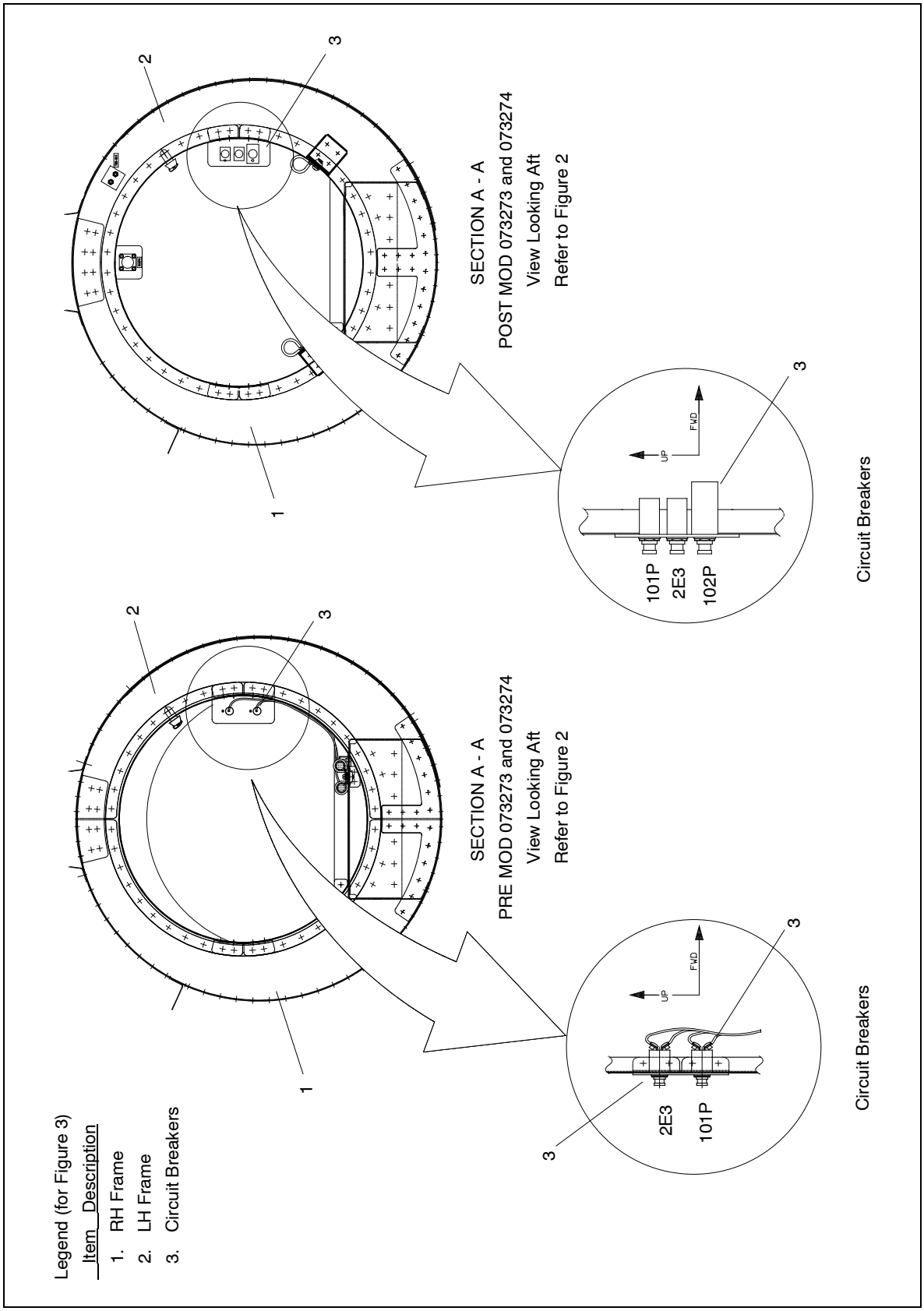
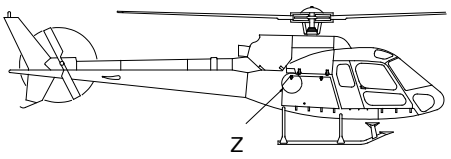
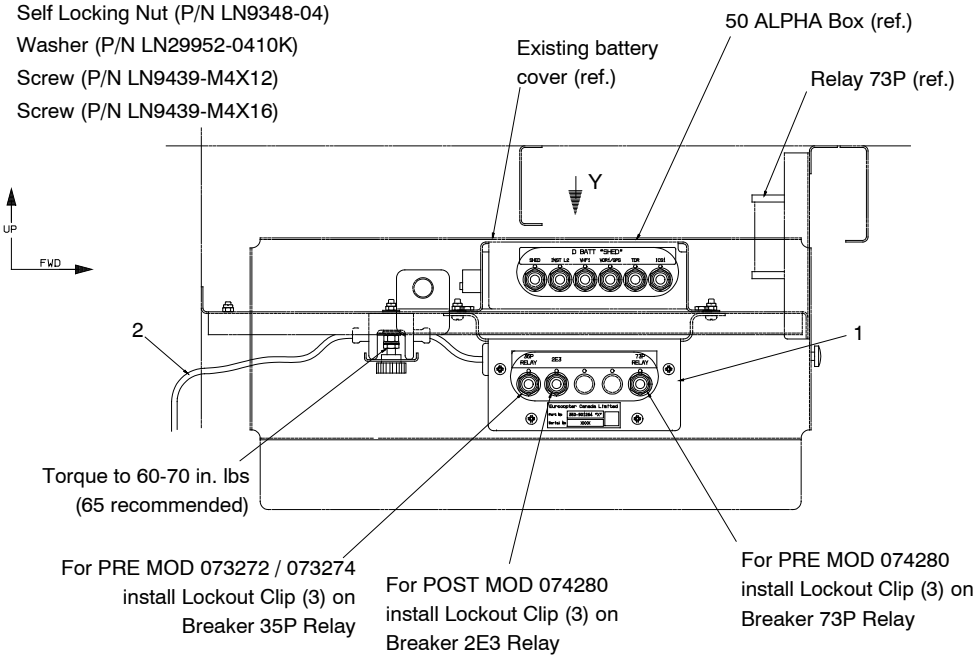


Figure 3 AS 350 Battery Compartment PRE and POST MOD 07-3273 and 07-3274 (detail views STN A 1817)

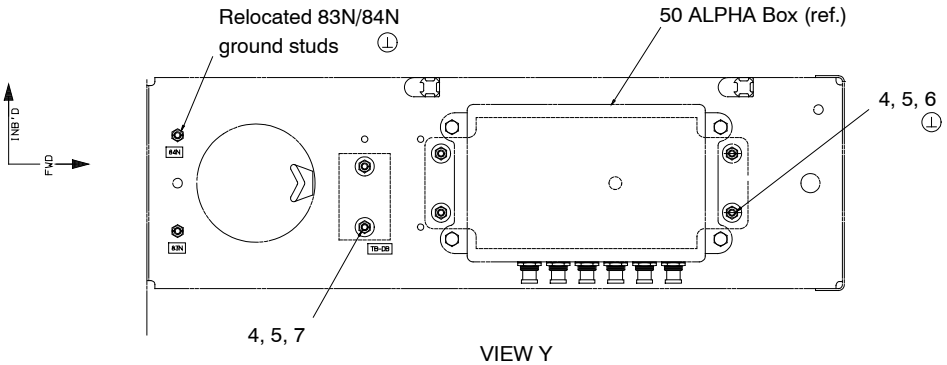
Legend (for Figure 4)

Item	Description
1.	Direct Battery Bus Circuit Breaker Box (part of Battery Harness, item 2)
2.	Battery Harness (P/N 350-901194-01)
3.	Lockout Clip
4.	Self Locking Nut (P/N LN9348-04)
5.	Washer (P/N LN29952-0410K)
6.	Screw (P/N LN9439-M4X12)
7.	Screw (P/N LN9439-M4X16)





VIEW Z
Looking inboard at the Direct Battery Bus Circuit Breaker Box



VIEW Y
Looking down on Shelf

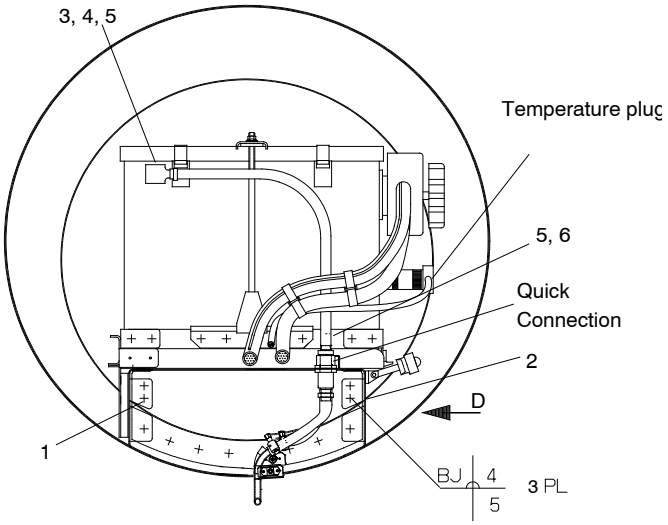
⏚ Contact between mating surfaces to be free from protective coating. Before installation, apply electrical joint compound.

NOTE:

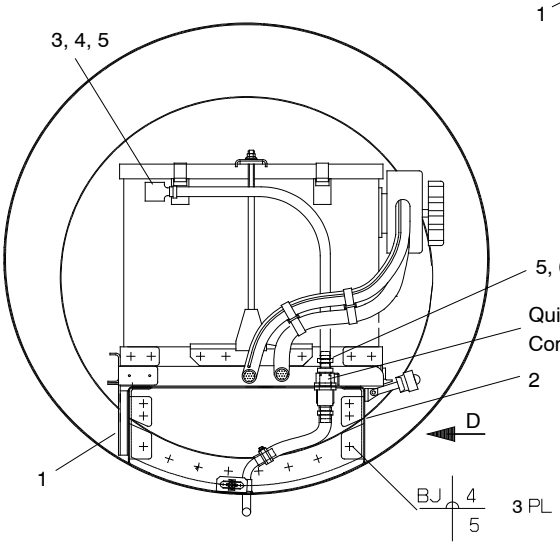
Figure 4 AS 350 Direct Battery Bus Circuit Breaker Box

Legend (for Figure 5)

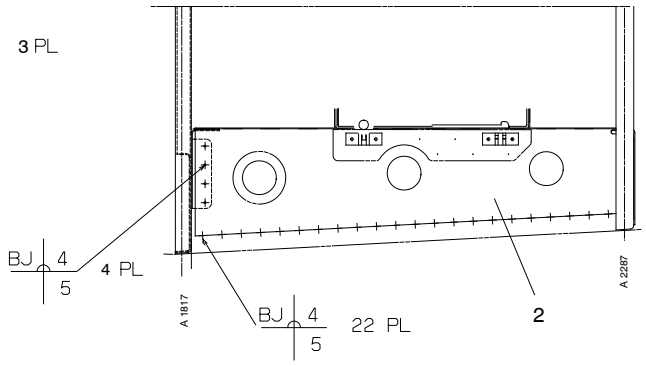
Item	Description
1.	RH Stiffener
2.	LH Stiffener
3.	Elbow
4.	Hose Clamp
5.	Tywrap
6.	Hose Barb



VIEW B
Nickel-Cadmium Battery
Refer to Figure 2



VIEW C
Lead-Acid Battery
Refer to Figure 2



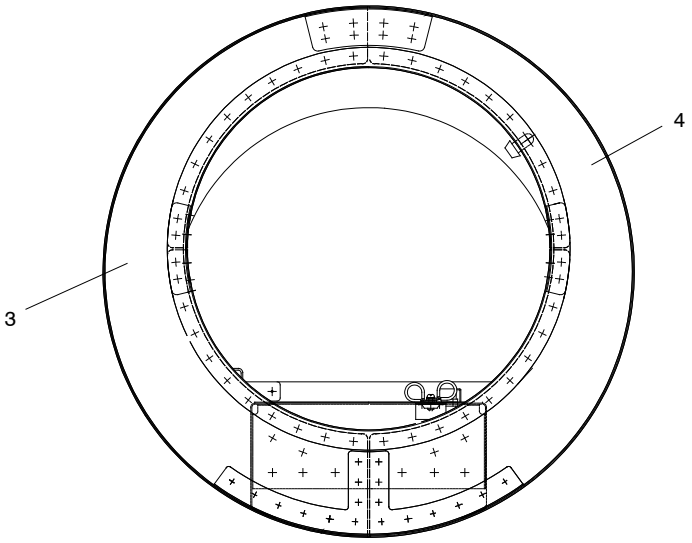
VIEW D
View of LH Stiffener tailboom skin removed

BJ	MS20470AD
CODE	RIVET TYPE
RIVET CODE PER NAS 523	

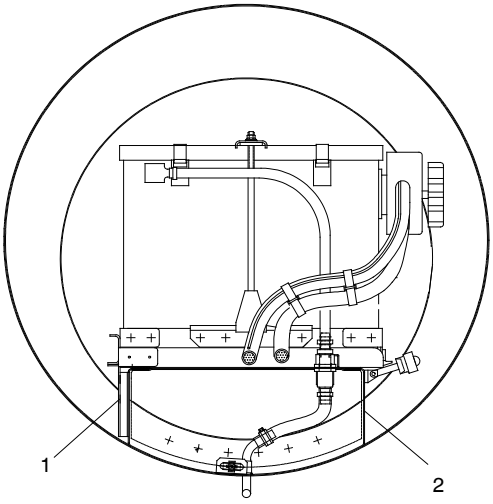
Figure 5 AS 350 Battery Compartment (detail views)

Legend (for Figure 6)

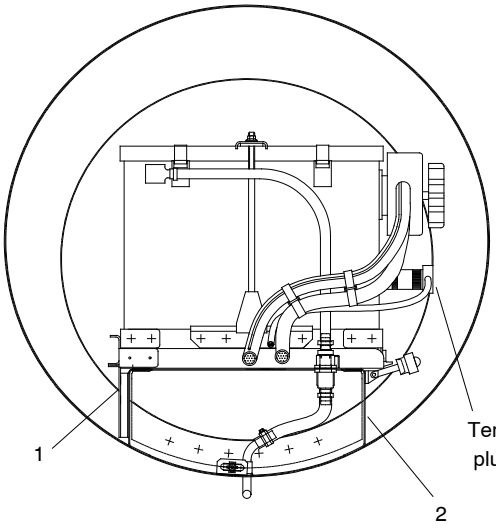
Item	Description
1.	RH Stiffener
2.	LH Stiffener
3.	RH Frame
4.	LH Frame



SECTION A - A
At STN A 1817 looking aft
Refer to Figure 2



VIEW C
Lead-Acid Battery
Refer to Figure 2



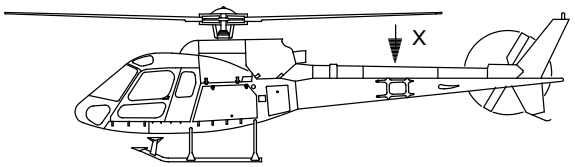
VIEW B
Nickel-Cadmium Battery
Refer to Figure 2

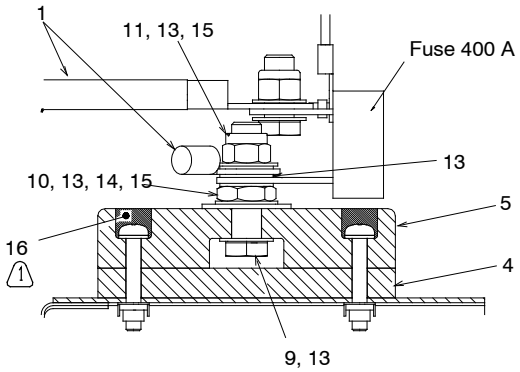
Note: For Section A-A, VIEW B and VIEW C locations refer to Figure 2.

Figure 6 AS 355 Battery Compartment (detail views STN A 1817)

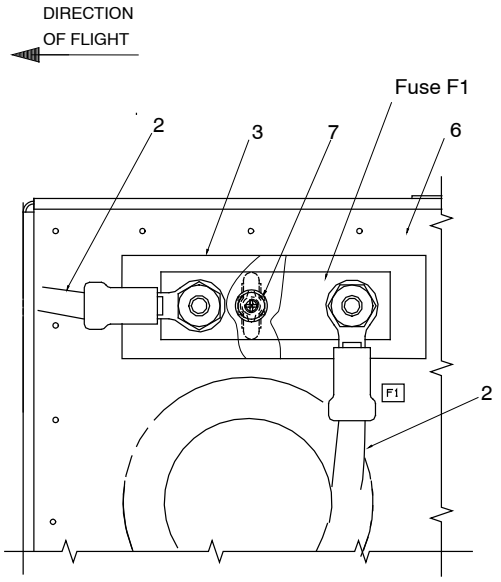
Legend (for Figure 7)

Item	Description
1.	Battery Harness (350-901194-03)
2.	Battery Harness (350-901194-02)
3.	Fuse Holder Cover
4.	Base Plate
5.	Terminal Support
6.	Base
7.	Screw (P/N LN9438-M5X20)
8.	Screw (P/N (LN9439-AM5X32)
9	Bolt (P/N DIN933M10X40A2-70)
10	Nut (P/N 22435CC100)
11.	Nut (P/N LN9448-10)
12.	Washer (P/N LN9016-05K)
13.	Washer (P/N LN9025-1010L)
14.	Washer (P/N LN29952-1015M)
15.	Spring Washer (P/N DIN137B10B3C)
16.	Sealant (P/N PR 1422-B2)

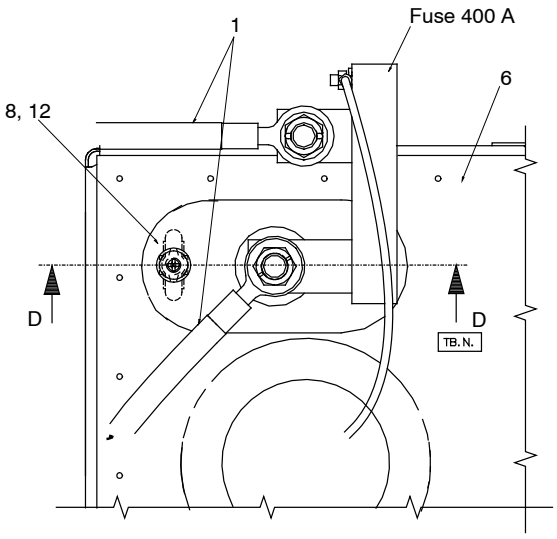




VIEW D - D
POST MOD 074836 - 02 Variant
View looking inboard



VIEW X
Looking down on Battery Tray
PRE MOD 074836 - 01 Variant
(Skin and battery removed for clarity)



VIEW X
Looking down on Battery Tray
POST MOD 074836 - 02 Variant
(Skin and battery removed for clarity)

1 Fill counterbore in terminal support (5) with sealant (16).

NOTE:

Figure 7 Terminal Supports on Battery Tray PRE and POST MOD 07-4836

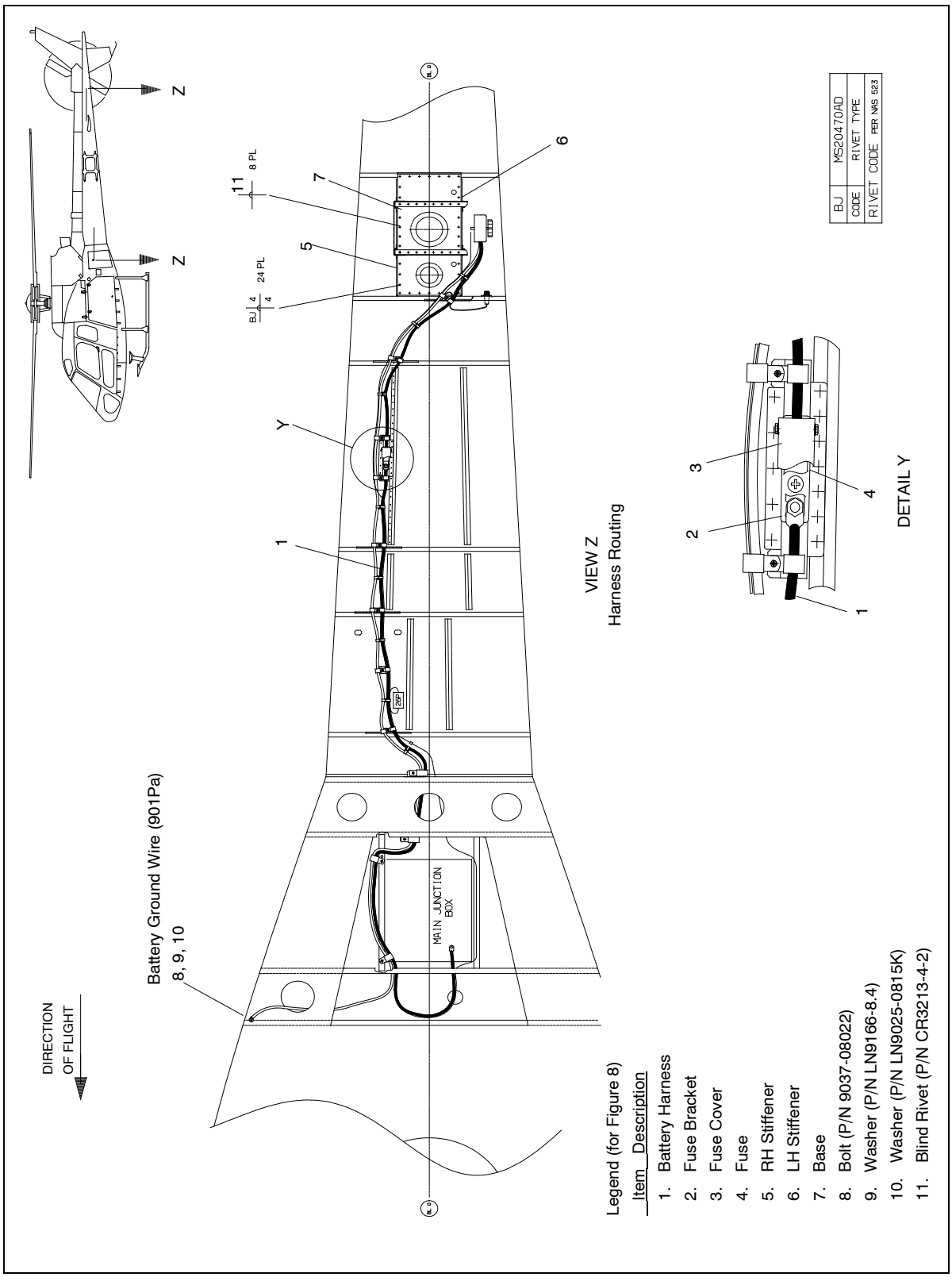


Figure 8 AS 350 Harness Routing – Tail Boom PRE MOD 07-3273 and 07-3274

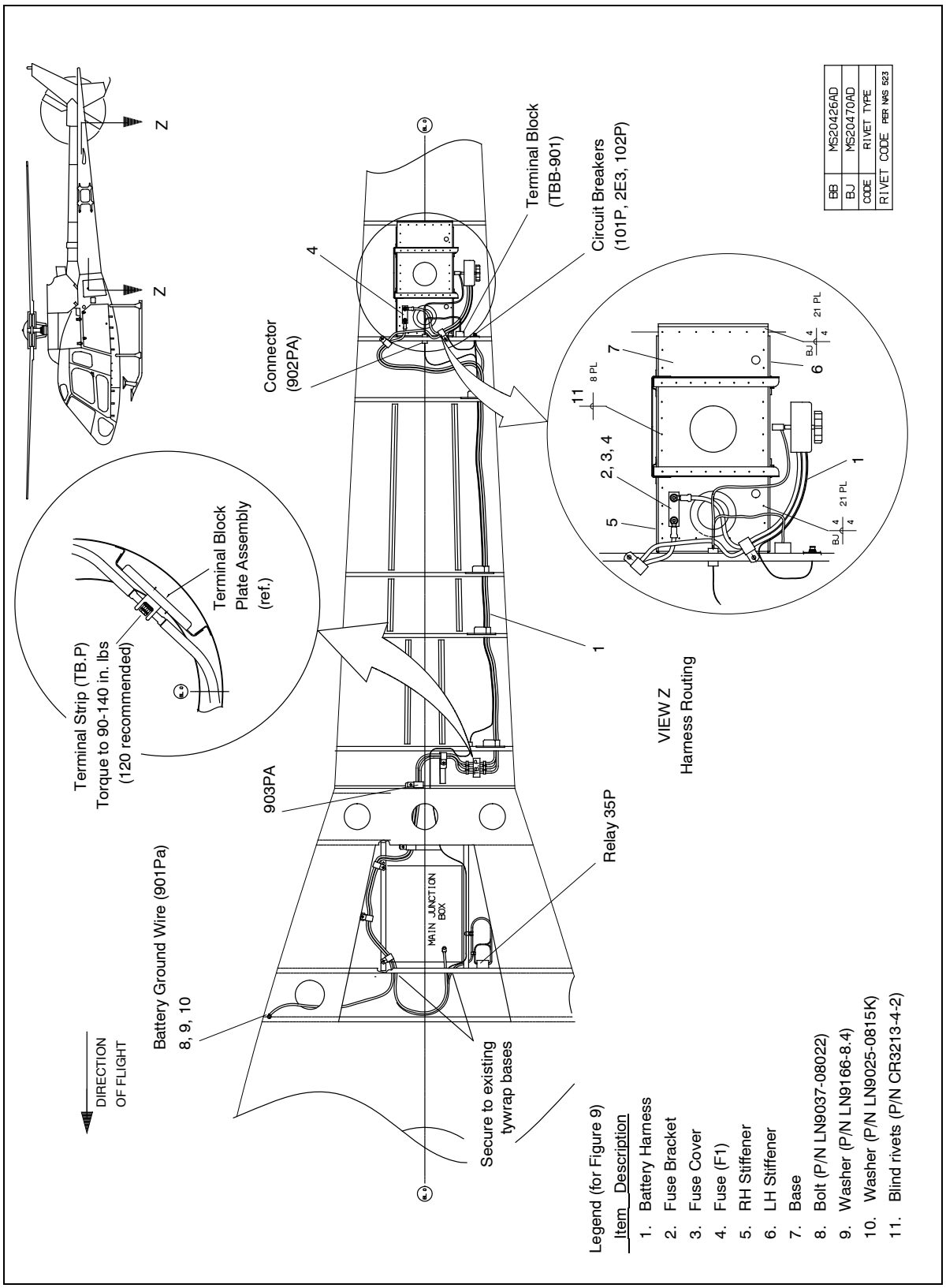


Figure 9 AS 350 Harness Routing – Tail Boom POST MOD 07-3273 and 07-3274

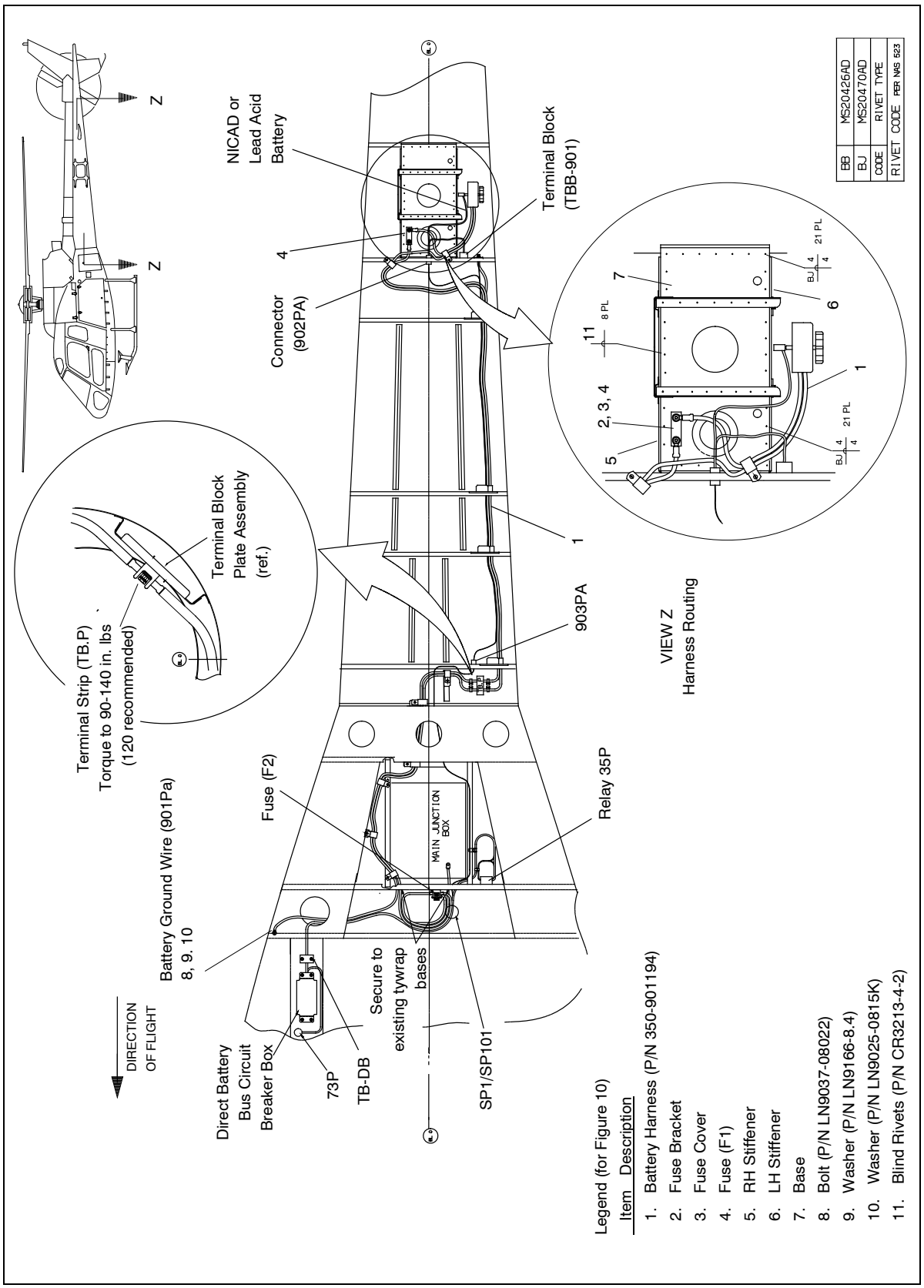
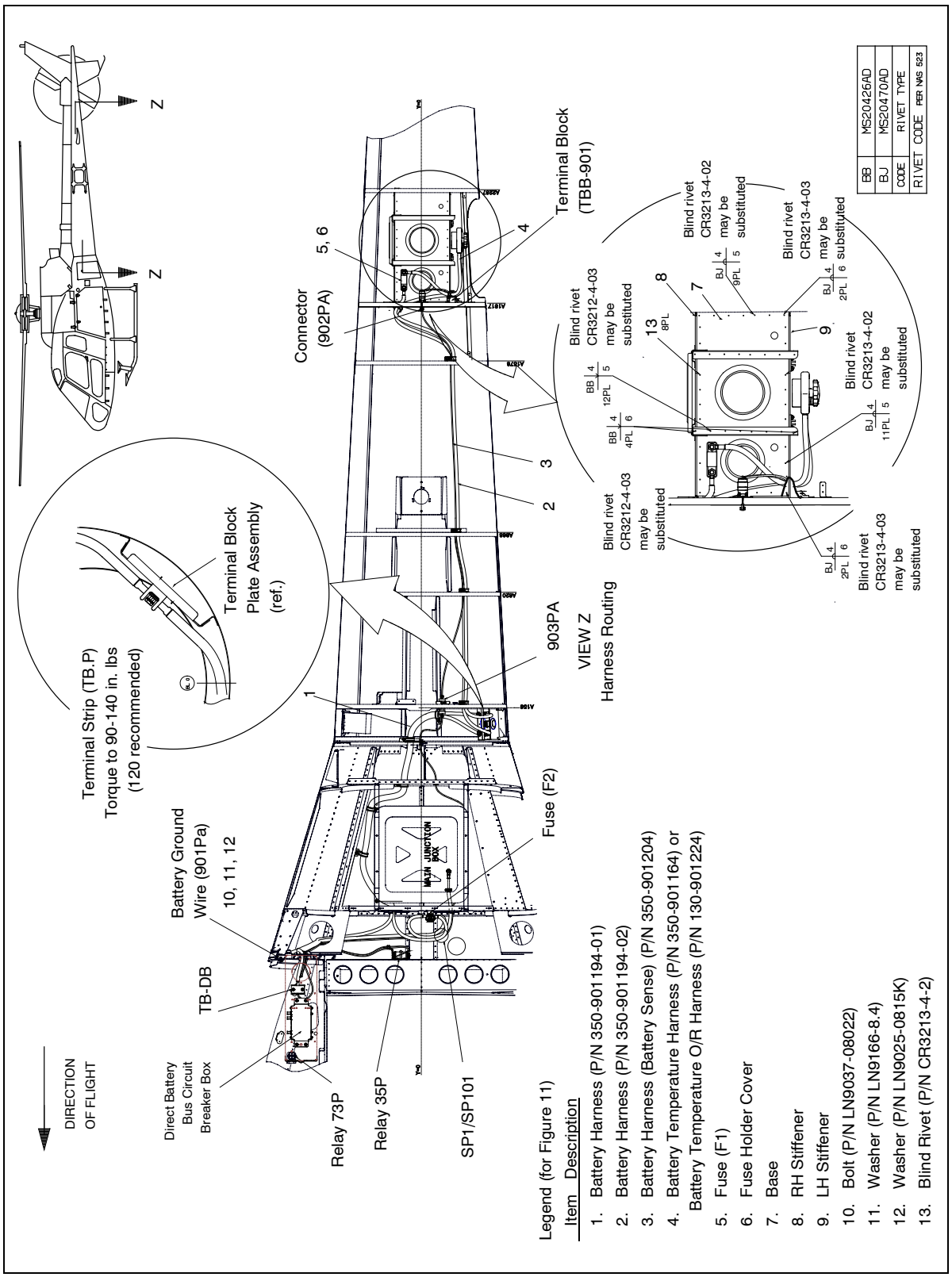


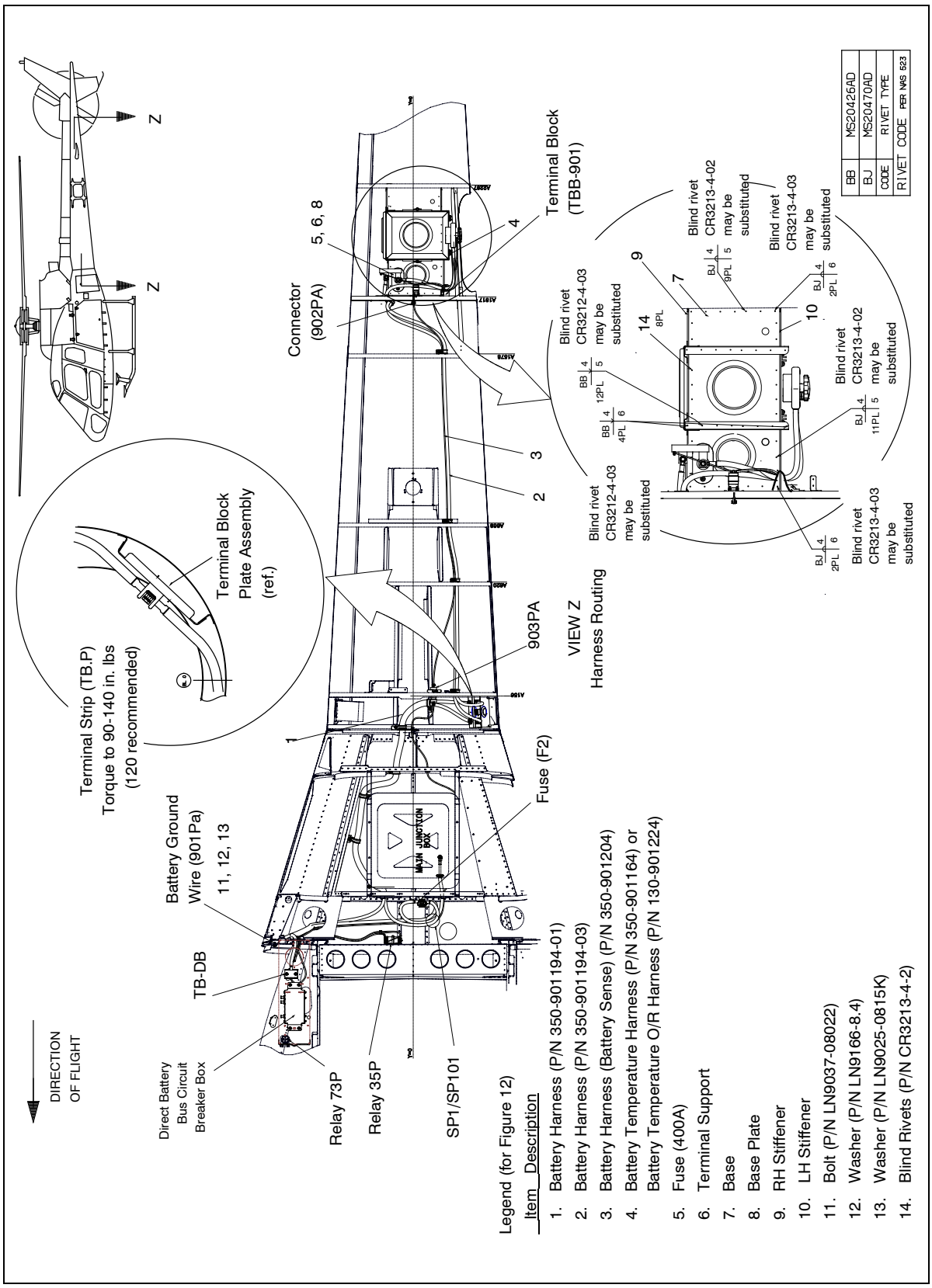
Figure 10 AS 350 Harness Routing – Tail Boom POST MOD 07-4280 (AS 350 B3)



Legend (for Figure 11)

Item	Description
1.	Battery Harness (P/N 350-901194-01)
2.	Battery Harness (P/N 350-901194-02)
3.	Battery Harness (Battery Sense) (P/N 350-901204)
4.	Battery Temperature Harness (P/N 350-901164) or Battery Temperature O/R Harness (P/N 130-901224)
5.	Fuse (F1)
6.	Fuse Holder Cover
7.	Base
8.	RH Stiffener
9.	LH Stiffener
10.	Bolt (P/N LN9037-08022)
11.	Washer (P/N LN9166-8.4)
12.	Washer (P/N LN9025-0815K)
13.	Blind Rivet (P/N CR3213-4-2)

Figure 11 AS 350 Harness Routing – Tail Boom PRE MOD 07-4836 – 01 Variant



Legend (for Figure 12)

Item	Description
1.	Battery Harness (P/N 350-901194-01)
2.	Battery Harness (P/N 350-901194-03)
3.	Battery Harness (Battery Sense) (P/N 350-901204)
4.	Battery Temperature Harness (P/N 350-901164) or Battery Temperature O/R Harness (P/N 130-901224)
5.	Fuse (400A)
6.	Terminal Support
7.	Base
8.	Base Plate
9.	RH Stiffener
10.	LH Stiffener
11.	Bolt (P/N LN9037-08022)
12.	Washer (P/N LN9166-8.4)
13.	Washer (P/N LN9025-0815K)
14.	Blind Rivets (P/N CR3213-4-2)

Figure 12 AS 350 Harness Routing – Tail Boom POST MOD 07-4836 – 02 Variant

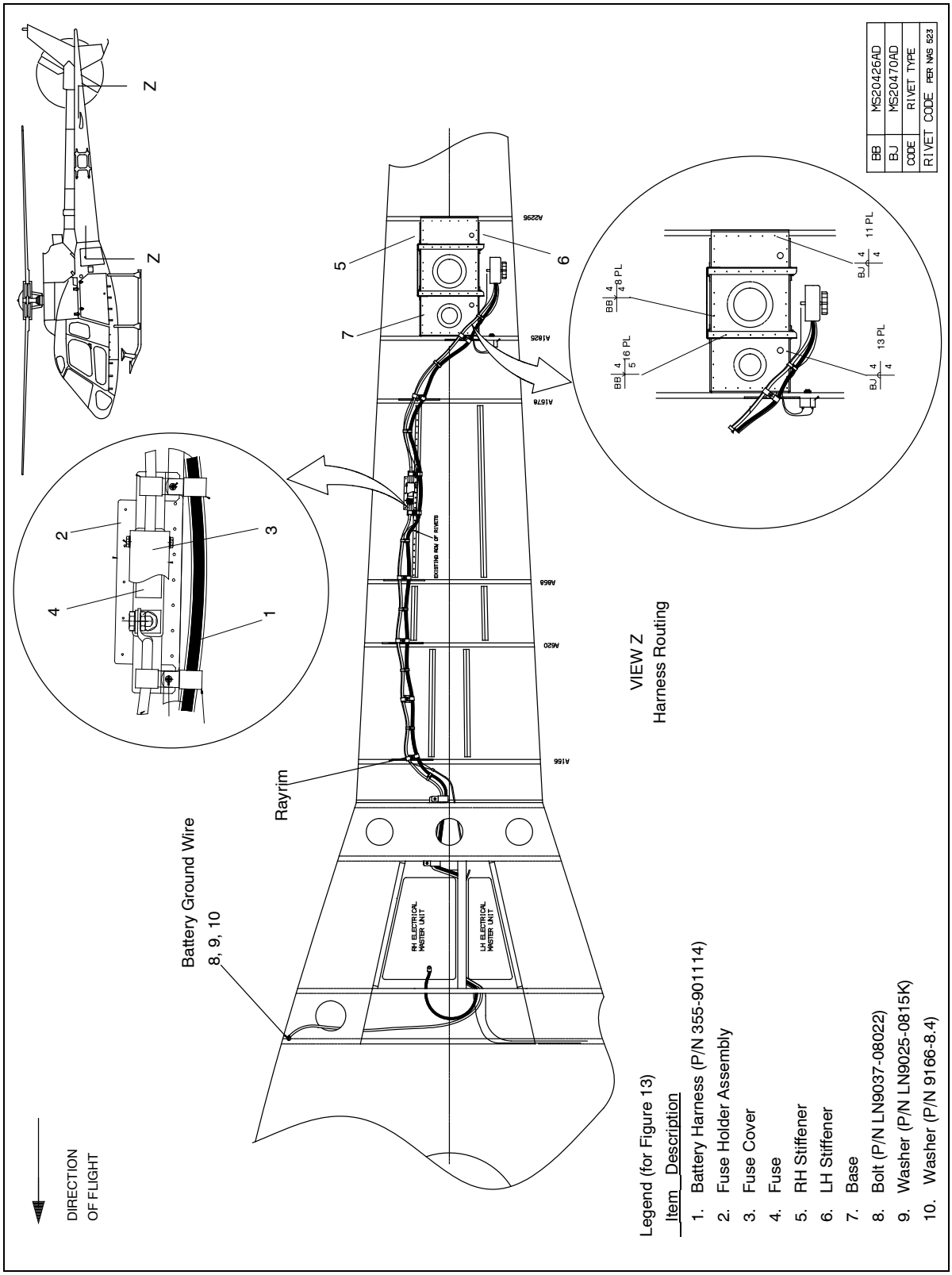
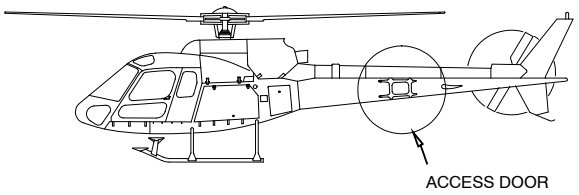
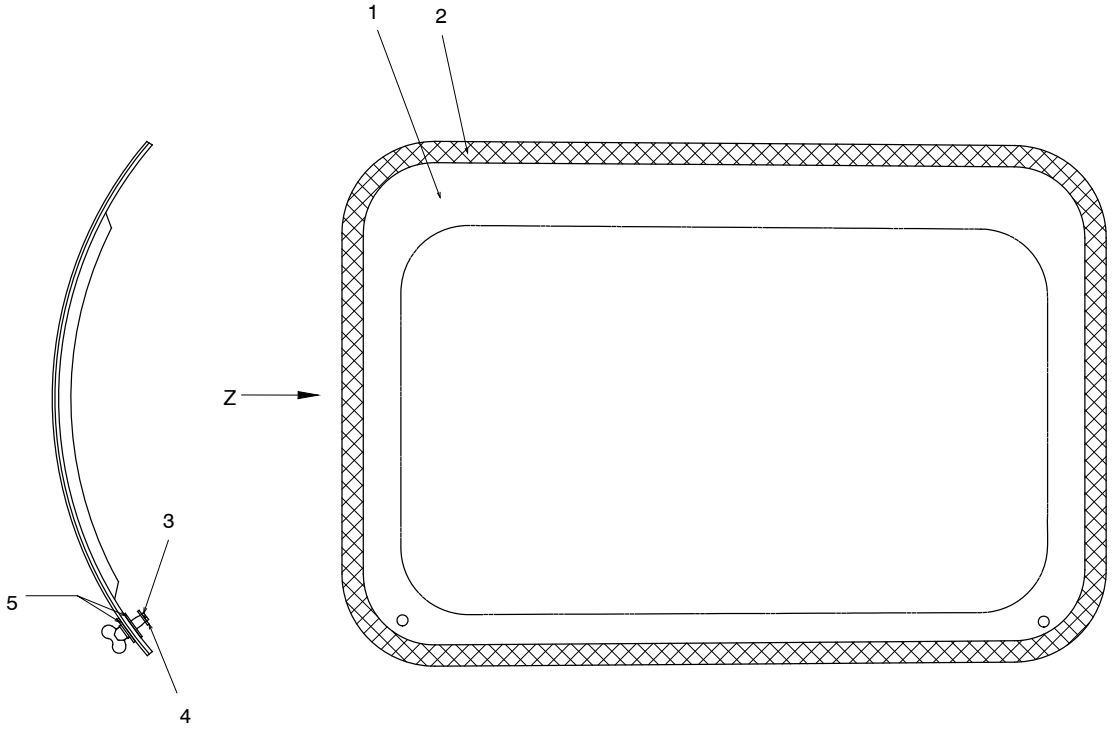


Figure 13 AS 355 Harness Routing – Tail Boom



Legend (for Figure 14)

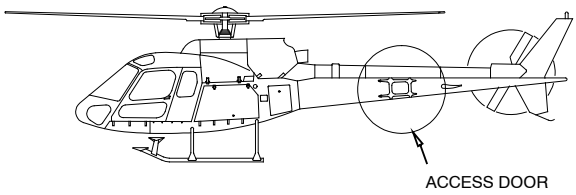
Item	Description
1.	Access Door
2.	Neoprene Closed Cell Sponge Self Adhesive (P/N SC41)
3.	1/4 Turn Fastener (P/N 2600-11SW)
4.	Retaining Rings (P/N 2600-SW2 or 55144-101)
5.	Plastic Washers (P/N 350A23-1030-20)



VIEW Z

INSIDE VIEW LOOKING OUTWARDS

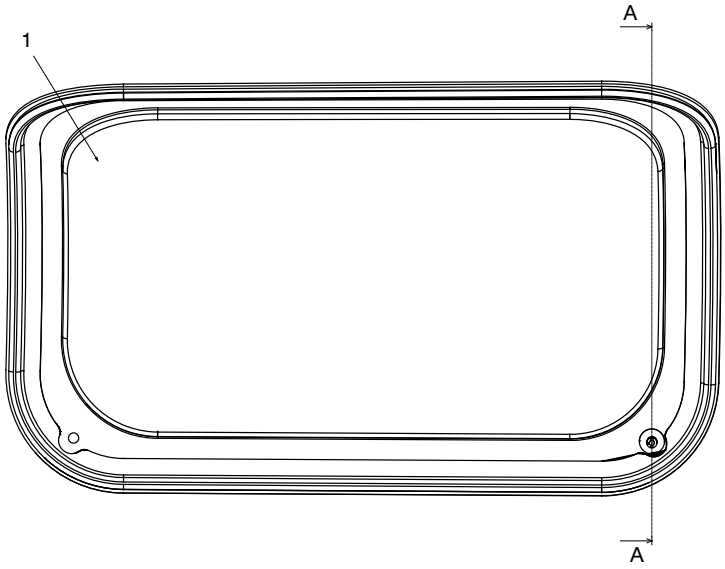
Figure 14 Access Door



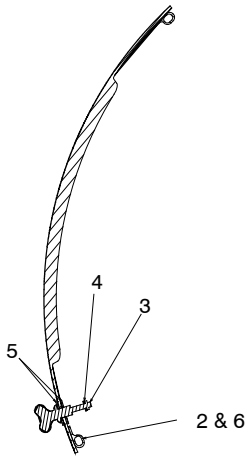
ACCESS DOOR

Legend (for Figure 15)

Item	Description
1.	Access Door
2.	P-Seal (P/N 1952)
3.	1/4 Turn Fastener (P/N 2600-12SW)
4.	Retaining Rings (P/N NSA 55144-003)
5.	Plastic Washers (P/N 350A23-1030-20)
6.	Adhesive (P/N 3M1300L or 3M1300)



FRONT VIEW



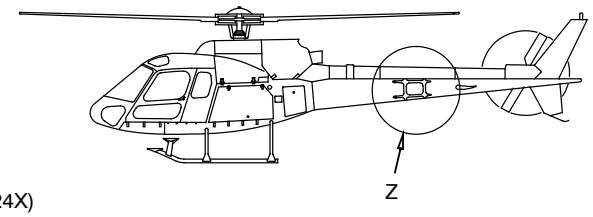
SECTION A - A

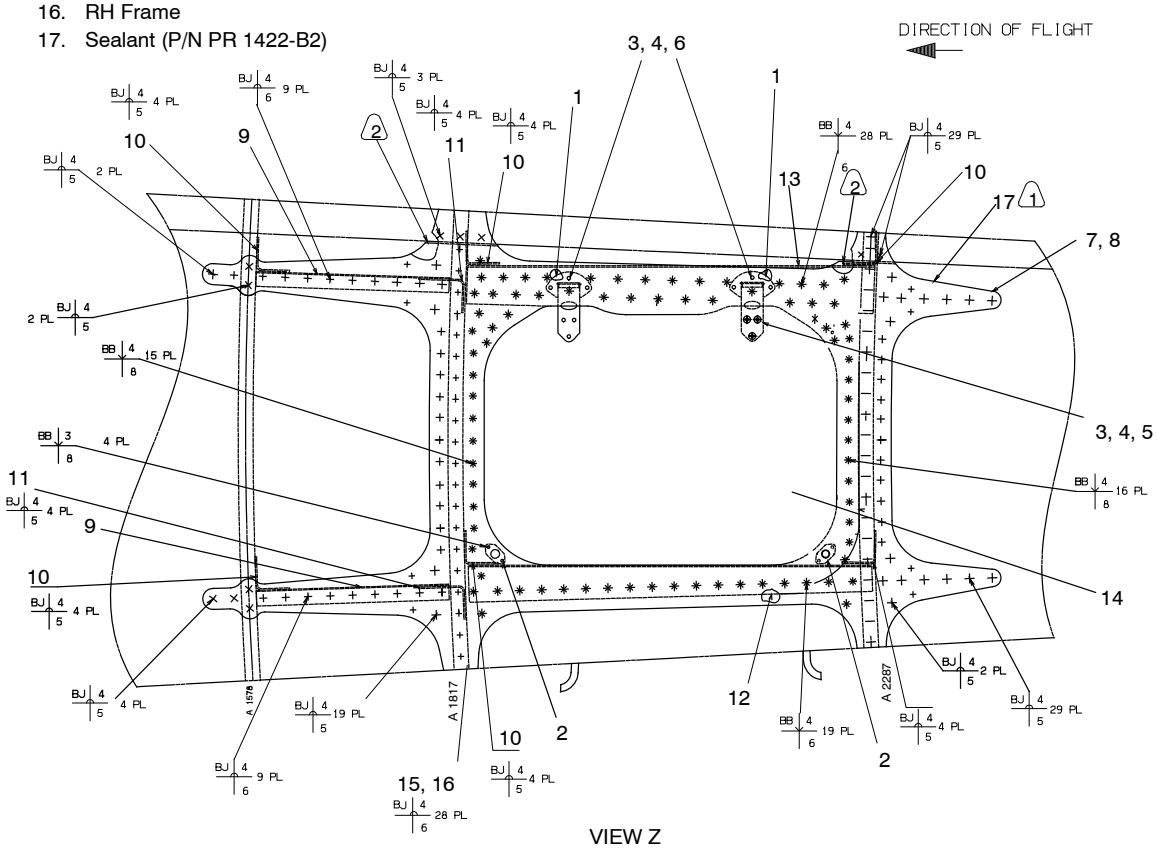
Figure 15 Access Door (continued)

Legend (for Figure 16)

Item Description

1. Hinge
2. Receptacle
3. Self Locking Nuts (P/N LN9348-04)
4. Washers (P/N LN9016-04K)
5. Screws (P/Ns ECS2365TK04S24X or A0164TK040S024X)
6. Screw (P/N LN9438-M4X16)
7. Outer Skin Doubler
8. Inner Skin Doubler
9. TC Stiffener
10. Stiffener Clip
11. Stiffener Clip
12. Lower Stiffener
13. Upper Stiffener
14. Access Door
15. LH Frame
16. RH Frame
17. Sealant (P/N PR 1422-B2)





VIEW Z

Showing Doublers, Access Door and Receptacles

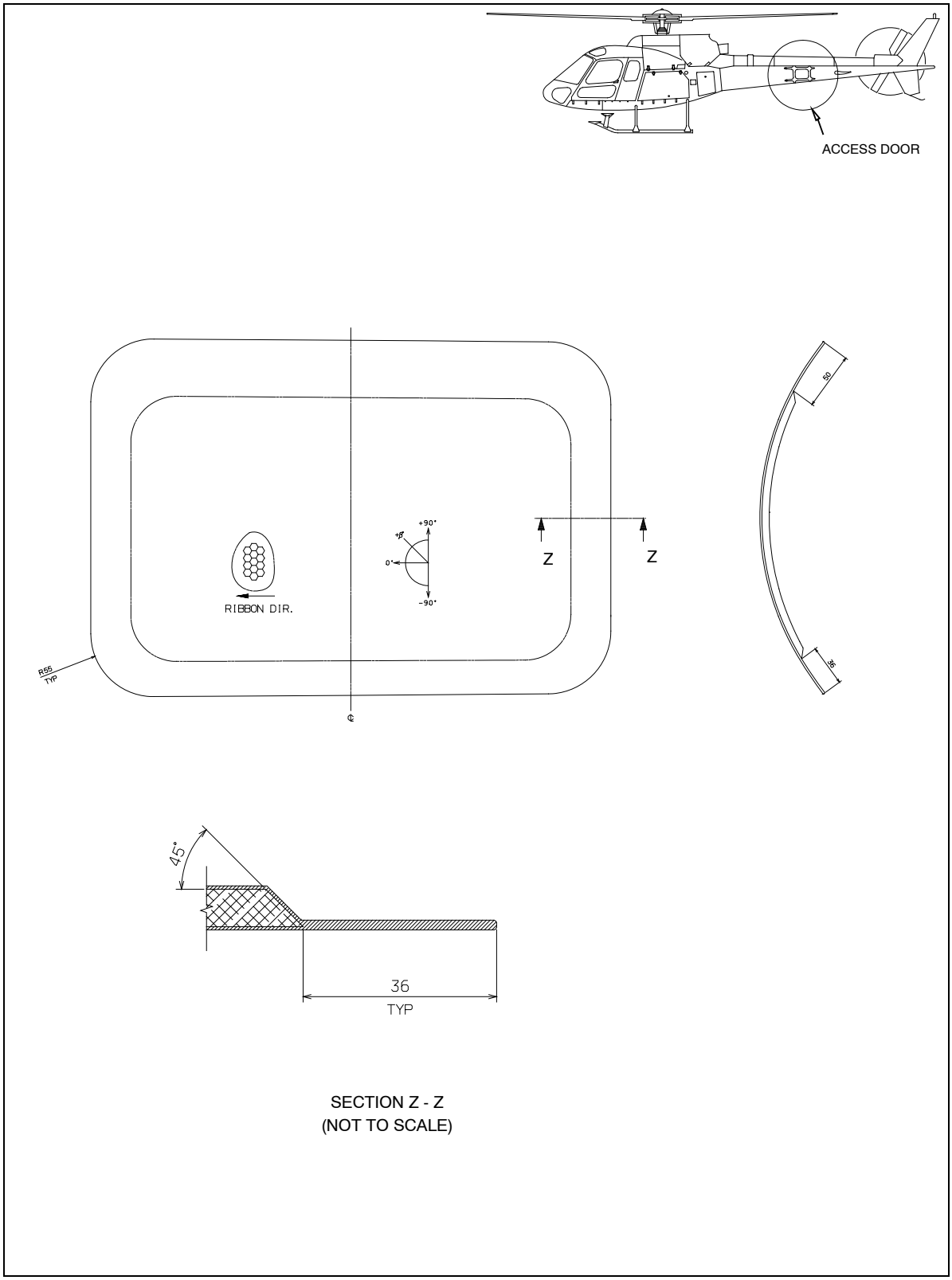
BB	MS20426AD
BJ	MS20470AD
CODE	RIVET TYPE
RIVET CODE	PER NAS 523

(2) Trim inner doubler (8) to clear upper tailboom skin.

(1) Seal seam using sealant (17).

NOTE:

Figure 16 Tail Boom Details



SECTION Z - Z
(NOT TO SCALE)

Figure 17 Access Door – Composite Layup

C. REFERENCES

DOCUMENT	DOCUMENT TITLE
AC-43.13 – 1B	Acceptable Methods, Techniques and Practices – Aircraft Inspection and Repair
AMM	Aircraft Maintenance Manual
CMM 24-30-71	Concorde, RG® Series Main Aircraft Battery, Component Maintenance Manual, Concorde, RG® Rev Q, Dec., 4 2018(or latest revision)
IPC	Illustrated Parts Catalog
IP-ECL-6 and H240I0221580	Installation Procedure, Battery Relocation
MET	Maintenance Manual
MOD 07-3254	Modification 07-3254: Arriel 281 Engine
MOD 07-3273	Modification 07-3273: new 30 ALP Control Unit (SMS)
MOD 07-3274	Modification 07-3274: fuses replaced by circuit breakers
MOD OP 3346	Modification Optional Equipment 3346: Dual Hydraulic System
MOD OP-3369	Modification 0P-3369: MTOW extended to 2,370 kg
MOD 07-4280	Modification 07-4280: SMS unit replaced by Multiple Control Unit
MOD 07-4836	Modification 07-4836: add fuse in the negative battery in order to protect aircraft against short circuit for the main battery
MTC	Standard Practices Manual
OMM 24-30-99	Operating and Maintenance Manual, SAFT, dated April 16, 2020 (or latest revision)

D. ABBREVIATION & DEFINITIONS

ABBREVIATION	DEFINITION
Acc'd	Accepted
AHCA	Airbus Helicopters Canada Limited
ALP	Alpha
App'd	Approved
ATA	Air Transportation Association
A/W	Airworthiness
BATT TEMP	Battery Temperature
CAR	Canadian Aviation Regulations
DAPM	Design Approval Procedures Manual
EASA	European Aviation Safety Agency
FAA	Federal Aviation Administration
FWD	Forward
ICA	Instructions for Continued Airworthiness
LH	Left-Hand
MTOW	Maximum Take-off Weight
P/N	Part Number
RH	Right-Hand
SMS	Safety Management System
STC	Supplemental Type Certificate
STN	Station
TCCA	Transport Canada Civil Aviationn
TGB	Tail Gear Box

E. UNITS OF MEASURE

ABBREVIATION / SYMBOL	UNIT OF MEASUREMENT
Ah	Ampere hour
D	Days
FH	Flight Hours
hrs	hours
in	inches
kg	kilograms
lb	pounds
m	meters
mm	millimeters
M	Months

2 AIRWORTHINESS LIMITATIONS

Canadian Approval
The Airworthiness Limitations section is approved by the Minister of Transport and specifies maintenance required by any applicable airworthiness or operating rule unless an alternative program has been approved by the Minister.

FAA Approval
The Airworthiness Limitations section is FAA approved and specifies inspections and other maintenance required under Secs. 43.16 and 91.403 of the Federal Aviation Regulations unless an alternative program has been FAA approved.

EASA Approval
The Airworthiness Limitations section is approved and variations must also be approved.

No Airworthiness Limitations associated with this installation.

3 CONTROL AND OPERATION

Apart from the following, control and operation of the aircraft remains unchanged:

If operating with the lead-acid battery, the aircraft “BATT TEMP” warning light is inoperative on the Instrument Panel.

For information on operating the Concorde RG® Series Main Aircraft Battery, refer to the Concorde Battery Corporation, Component Maintenance Manual, Document Number 5-0171, CMM 24-30-71, Rev. Q, Dec., 4, 2018 (or latest version).

For information on operating the nickel-cadmium battery SAFT 2376 Series, refer to the SAFT Operating and Maintenance Manual, Document Number OMM 24-30-99, February 14, 2024 (or latest version).

4 INSPECTION SCHEDULE AND MAINTENANCE ACTION

Refer to Section 8 if removing or replacing any parts.

CAUTION: PRIOR TO WORKING ON THE BATTERY OR BATTERY CIRCUIT ENSURE THAT THE AIRCRAFT ELECTRICAL SYSTEM IS NOT ENERGIZED.

NOTE: For battery inspection schedule and functional test refer to the appropriate manufacturer's instructions. Refer to Section 5 of this document for more information.

For additional information on scheduled inspections and troubleshooting, refer to the Concorde Battery Corporation RG® Series, Instructions for Continued Airworthiness, Concorde Valve Regulated Lead-Acid Main Battery, Document Number 5-0171, CMM 24-30-71, Rev. Q, dated Dec., 4, 2018 (or latest version). Refer to Section 5 of this document for more information.

If operating with the nickel-cadmium battery SAFT 2376 Series, refer to the SAFT Operating and Maintenance Manual, Document Number OMM 24-30-99, dated February 14, 2024 (or latest version). Refer to Section 5 of this document for more information.

NOTE: The latest revision of the Concorde and Saft Operations Manuals are available for download on their websites. Please refer to Section 5 of this document for more information.

NOTE: Use torque per MTC 20-02-05-404, unless otherwise specified.

4.1. INSPECTION SCHEDULE

4.1.1. Before the first flight of each day:

ITEM	INSPECTION OR MAINTENANCE WORK	CORRECTIVE ACTION
A	- Visually inspect battery connector, item 11 in Figure 2, for: a. security	 a. Secure as required.

Table 1 Inspection Schedule and Maintenance Action
Before the first flight of each day

NOTE: The “Before the first flight of each day” task can be carried out by suitably trained pilot or maintenance personnel.

4.1.2. SAFT Battery: Periodical check. Check at specific intervals according to aircraft use or every 3 M:

ITEM	INSPECTION OR MAINTENANCE WORK	CORRECTIVE ACTION
A	- Perform periodical check in accordance with the battery manufacturers recommendations.	For more information refer to the SAFT Nickel-Cadmium Aircraft Batteries Operating and Maintenance Manual.

Table 2 Inspection Schedule and Maintenance Action
SAFT Battery: Periodical check. Check at specific intervals according to aircraft use or every 3 M

4 INSPECTION SCHEDULE AND MAINTENANCE ACTION (continued)

4.1. INSPECTION SCHEDULE (continued)

4.1.3. SAFT Battery: Regular check and general overhaul. Check according to aircraft use or after a maximum of 12 M:

ITEM	INSPECTION OR MAINTENANCE WORK	CORRECTIVE ACTION
A	- Perform regular check in accordance with the battery manufacturers recommendations. - Perform general overhaul in accordance with the battery manufacturer's recommendations.	For more information refer to the SAFT Nickel-Cadmium Aircraft Batteries Operating and Maintenance Manual.

Table 3 Inspection Schedule and Maintenance Action
SAFT Battery: Regular check and general check.
Check according to aircraft use or after a maximum of 12 M

4.1.4. CONCORDE Battery if operating less that 1000 hrs per year after initial installation: Check at 12 M after initial installation (±30 D):

ITEM	INSPECTION OR MAINTENANCE WORK	CORRECTIVE ACTION
A	- Perform routine maintenance in accordance with the battery manufacturers recommendations. Operating less than 1000 hrs per year.	Perform capacity test and maintain as per Concorde Battery RG® Series Main Aircraft Battery Component Maintenance Manual. Refer to Testing and Fault Isolation paragraph 1A1.

Table 4 Inspection Schedule and Maintenance Action
CONCORDE Battery if operating less that 1000 hrs per year after initial installation:
Check at 12 M after initial installation (±30 D)**:**

4.1.5. CONCORDE Battery if operating 1000 hrs per year or more after initial installation (±100 hrs):

ITEM	INSPECTION OR MAINTENANCE WORK	CORRECTIVE ACTION
A	- Perform routine maintenance in accordance with the battery manufacturers recommendations. Operating more than 1000 hrs per year.	Perform capacity test and maintain as per Concorde Battery RG® Series Main Aircraft Battery Component Maintenance Manual. Refer to Testing and Fault Isolation paragraph 1A2.

Table 5 Inspection Schedule and Maintenance Action
CONCORDE Battery: Initial check at 1000 hrs per year or more after initial installation (±100 hrs)

4 INSPECTION SCHEDULE AND MAINTENANCE ACTION (continued)

4.1. INSPECTION SCHEDULE (continued)

4.1.6. Every 150 FH or 12 M (Margins: 15 FH or 36 D) to coincide with the 150 FH or 12 M helicopter inspection, whichever occurs first:

ITEM	INSPECTION OR MAINTENANCE WORK	CORRECTIVE ACTION
A	- Visually inspect battery connector, item 11, in Figure 2 for: a. general condition	 a. If damaged, contact AHCA for replacement parts.
B	- Visually check battery compartment, view shown in Figure 2 for: a. spilled acid or traces of alkaline electrolyte	 a. Remove battery and neutralize spills as required in accordance with AC43.13-1B, Chapter 11-20. Also refer to AS 350 (except B2/B3) and AS 355, MET, Chapter 24-30-00-201. For the AS 350 B2/B3 refer to AMM Chapter 24-33-00, 2-1.
C	- Check FWD vent hose, item 3 and AFT vent hose, item 4, in Figure 2 for: a. clogging and kinking b. cracking	 a. Clean and adjust as required. b. No cracking is allowed. If cracking is found, contact AHCA for replacement parts.
D	- Visually inspect battery tray assembly, item 6, in Figure 2 for: a. cracks and deformation b. corrosion c. scoring	 a. No cracks or deformation are allowed. If cracks or deformation are found, contact AHCA for replacement parts. b. No corrosion is allowed. If corrosion is found, contact AHCA for replacement parts. c. No scoring is allowed. If scoring is found, contact AHCA for replacement parts.
E	- Check battery tray assembly, item 6, in Figure 2 for: a. security	 a. Re-tighten as required.

Table 6 Inspection Schedule and Maintenance Action
Every 150 FH or 12 M (Margins: 15 FH or 36 D) to coincide with the 150 FH or 12 M helicopter inspection, whichever occurs first
(continued on following page)

4 **INSPECTION SCHEDULE AND MAINTENANCE ACTION** (continued)

4.1 INSPECTION SCHEDULE (continued)

4.1.6. Every 150 FH or 12 M (Margins: 15 FH or 36 D) to coincide with the 150 FH or 12 M helicopter inspection, whichever occurs first:

ITEM	INSPECTION OR MAINTENANCE WORK	CORRECTIVE ACTION
F	- Visually inspect U-plate, item 7, and battery clamp assembly, item 9, in Figure 2 for:	
	a. cracks or deformation	a. No cracks or deformation are allowed. If cracks or deformation are found, contact AHCA for replacement parts.
	b. corrosion	b. No corrosion is allowed. If corrosion is found, contact AHCA for replacement parts.
	c. scoring	c. No scoring is allowed. If scoring is found, contact AHCA for replacement parts.
G	- Visually inspect mounting hardware, self-locking nuts, item 8, and FWD and AFT locking knobs, item 5, in Figure 2 for:	
	a. security	a. Re-tighten as required.
	b. corrosion	b. No corrosion is allowed. If corrosion is found, contact AHCA for replacement parts.
	c. scoring	c. No scoring is allowed. If scoring is found, contact AHCA for replacement parts.
H	- Visually inspect access door mounting hardware, items 12, 13, 14, 15, and 16 in Figure 2 for:	
	a. security	a. Re-tighten as required.
I	- Visually inspect circuit breakers (PRE & POST MOD 07-3273 and 3274), item 3, in Figure 3, for:	
	a. secure mounting	a. Secure as required.
	b. general condition (physical damage)	b. If physical damage is evident, contact AHCA for replacement parts.
J	- Check mounting hardware, items 4, 5 and 6 securing the 50 ALPHA Box in Figure 4, for:	
	a. Secure mounting	a. Secure as required.

Table 6 Inspection Schedule and Maintenance Action
Every 150 FH or 12 M (Margins: 15 FH or 36 D) to coincide with the 150 FH or 12 M helicopter inspection, whichever occurs first
(continued on following page)

4 **INSPECTION SCHEDULE AND MAINTENANCE ACTION** (continued)

4.1 INSPECTION SCHEDULE (continued)

4.1.6. Every 150 FH or 12 M (Margins: 15 FH or 36 D) to coincide with the 150 FH or 12 M helicopter inspection, whichever occurs first:

ITEM	INSPECTION OR MAINTENANCE WORK	CORRECTIVE ACTION
K	- Visually inspect Battery Harness, item 1 in Figures 8, 9 10 and 13 and items 1, 2, and 3 in Figures 11 and 12 for:	
	a. cracks, fraying, burns and chaffing	a. If cracks, fraying, burns or chaffing are evident, contact AHCA for replacement harness and refer to IP-ECL-6 and IP H240I0221580 for installation instructions.
I	b. loose connections	b. Re-tighten as required.
	c. security	c. Re-tighten as required.
L	- Visually inspect access door seal, item 2, in Figure 14 and 15, for:	
	a. cuts or cracking, debonding or loss of elasticity	a. If cuts or cracking, debonding or loss of elasticity is evident, contact AHCA for replacement seal.
M	- Check access door latching hardware, items 3, 4 and 5 in Figure 14 and 15, for:	
	a. proper latching b. condition	a. Check for freedom of movement. b. Replace as required.
N	- Visually inspect outer skin doubler, item 7 in Figure 16, and the tail boom skin in the area of the cutout for:	
	a. scratches, cracks, perforation, corrosion or distortion	a. If scratches, cracks, perforation, corrosion or diistortion is found within tolerance, repairs may be accomplished with AMM, Chapter 53-31-00, 8-1. For scratches, cracks, perforation, corrosion or distortion found outside tolerance, contact AHCA for replacement parts. **NOTE:** If removal of oouter skin doubler (7) is required, inspect inner skin doubler (8) as well.

Table 6 Inspection Schedule and Maintenance Action
Every 150 FH or 12 M (Margins: 15 FH or 36 D) to coincide with the 150 FH or 12 M helicopter inspection, whichever ooccurs first
(continued on following page)

4 **INSPECTION SCHEDULE AND MAINTENANCE ACTION** (continued)

4.1 INSPECTION SCHEDULE (continued)

4.1.6. Every 150 FH or 12 M (Margins: 15 FH or 36 D) to coincide with the 150 FH or 12 M M helicopter inspection, whichever occurs first:

ITEM	INSPECTION OR MAINTENANCE WORK	CORRECTIVE ACTION
O	- Visually inspect access ddoor, item 1, in Figures 14 and 15 for: a. cracking, depression, delamination or holes	 a. No cracking, depression, delamination or holes allowed. Minor repairs may be accomplished in accordance with MTC, Chapter 20-03-07-101 or AC-43.13-1B, Chapter 3, Section 3.1 to 3.4. For major repairs, contact AHCA for repair information.
P	- Check placards and markings (refer to Section 10) for: a. legibility b. secure mounting	 a. If placard has become illegible, contact AHCA for replacement parts b. Secure or reattach placards as required.

Table 6 Inspection Schedule and Maintenance Action
Every 150 FH or 12 M (Margins: 15 FH or 36D) to coincide with the 150 FHH or 12 M helicopter inspection, whichever occurs first

4.1.7. Every 600 FH or 24 M (Margins: 60 FH or 73 D) to coincide with the 600 FH or 24 M helicopter inspection, whichever occurs first:
or
If you are operating AS 350 B3 only:
Every 750 FH or 24 M (Margins: 75 FH or 73 D) to coincide with the 750 FH or 24 M helicopter inspection, whichever occurs first:

ITEM	INSPECTION OR MAINTENANCE WORK	CORRECTIVE ACTION
A	- If nickel-cadmium, Saft 2376 battery type is installed, VIEW B in Figure 5, check temperature sensor harness connection and condition, check “BATT TEMP” warning light on warning caution panel.	Check battery temperature sensor in accordance with AS 350 (except AS 350 B2/B3) MET, Chapter 24-30-00-502. For AS 350 B2/B3 refer to AMM, 24-33-00, 5-1.
B	- Visually inspect battery ground wire hardware in rear cargo hold, Figures 8, 9, 10 11, 12 and 13 for: a. security	 a. Secure as required.

Table 7 Inspection Schedule and Maintenance Action
Every 600 FH or 24 M (Margins: 60 FH or 73 D) to coincide with the 600 FH or 24 M helicopter inspection, whichever occurs first
or
Every 750 FH or 24 M (Margins: 75 FH or 73 D) to coincide with the 750 FH or 24 M helicopter inspection, whichever ooccurs first

5 REPLACEMENT COMPONENTS AND REPAIR / OVERHAUL INFORMATION

Contact AHCA for replacement component parts. No overhaul information required for this installation.

For information contact Airbus Helicopter Customer Support Representatives:

Email: hcaresupport.canada@airbus.com
After Hours AOG Support: 1-800-267-4999
Visit our website at www.airbushelicopters.ca

CAUTION **DO NOT REPAIR OR OVERHAUL THE CONCORDE OR SAFT BATTERY. CONTACT CONCORDE OR SAFT FOR INFORMATION ON COMPONENT MAINTENANCE OR REPAIR.**

NOTE **CONCORDE RECOMMENDS REPLACING BATTERY AFTER FOUR YEARS SERVICE LIFE.**

NOTE: The latest revision of the Concorde and Saft Operations Manuals are available for download on their websites.
Please refer to information below:

Contact Concorde RG 2390 ER Series battery:

Concorde Battery Corporation
2009 San Bernardino Road
West Covina, CA 91790 USA
Telephone (626) 813-1234
www.concordebattery.com

Contact SAFT 2376-1 Series battery:

Saft
12, rue Sadi Carnot
93170 Bagnolet - France
Telephone: +33 (0) 1 49 93 19 18
www.saft4u.saft.com/en/user/login

6 TROUBLESHOOTING

If battery fails to perform to specification, refer to the Concorde, RG® Series Main Aircraft Battery. Document Number 5-0171, ATA 24-30-71, Rev. Q, Dec., 4, 2018 (or latest version), or the SAFT Operating and Maintenance Manual, Document Number ATA 24-30-99, February 14, 2024 (or latest version).

For electrical system troubleshooting for the AS 350, refer to Figures 18 to 23, Battery Relocation, Wiring Diagram.

For electrical system troubleshooting for the AS 355, refer to Figure 24, Battery Relocation, Wiring Diagram.

ITEM	TROUBLE / SYMPTOM	PROBABLE CAUSE	CORRECTIVE ACTION
1	“BATT TEMP” warning light on the Warning Caution Panel illuminates during flight.	Faulty battery	Refer to Component Maintenance Manual for battery troubleshooting information.

Table 8 Troubleshooting Guide

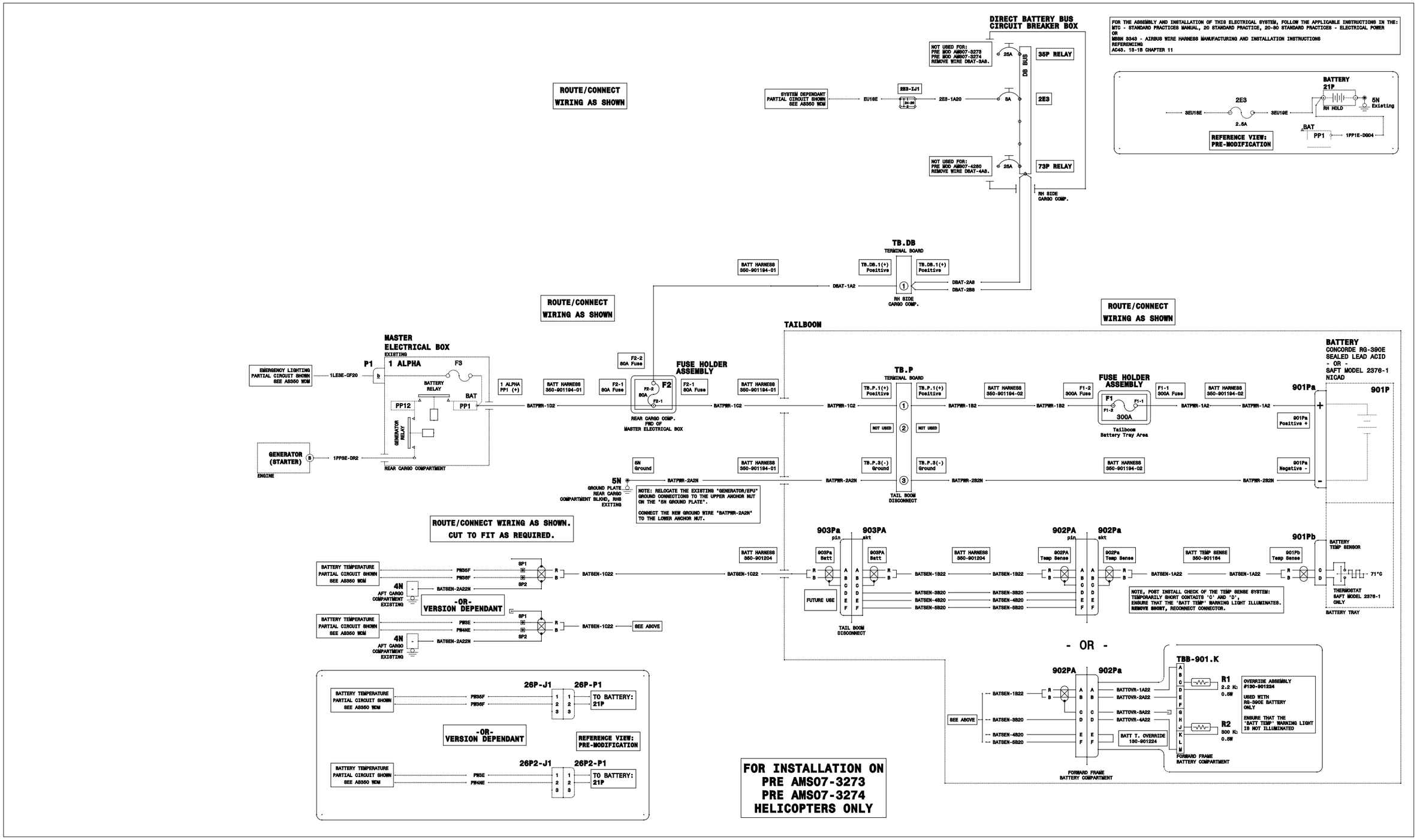


Figure 18 AS 350 Battery Relocation, Page 1 of 6, Wiring Diagram

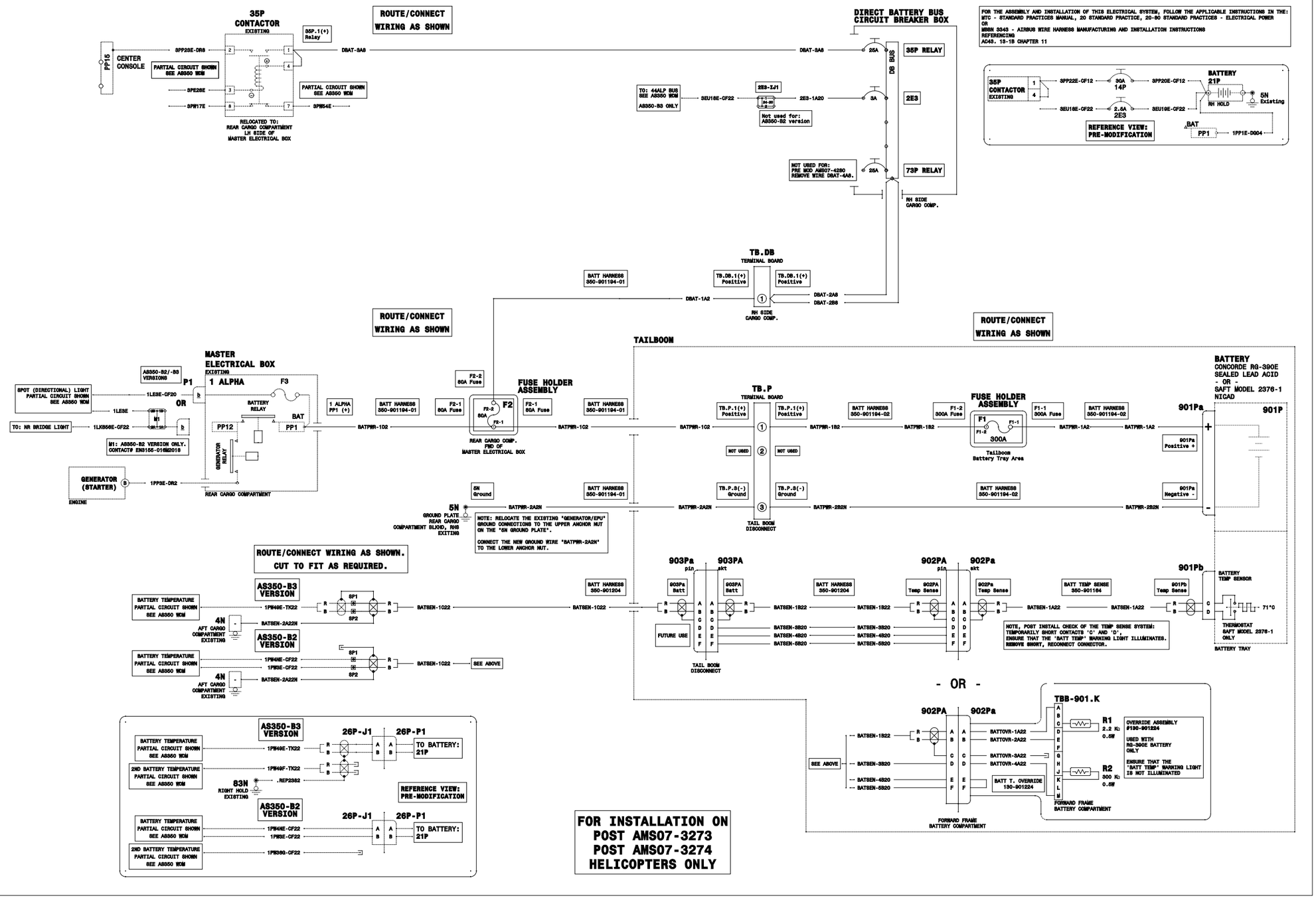
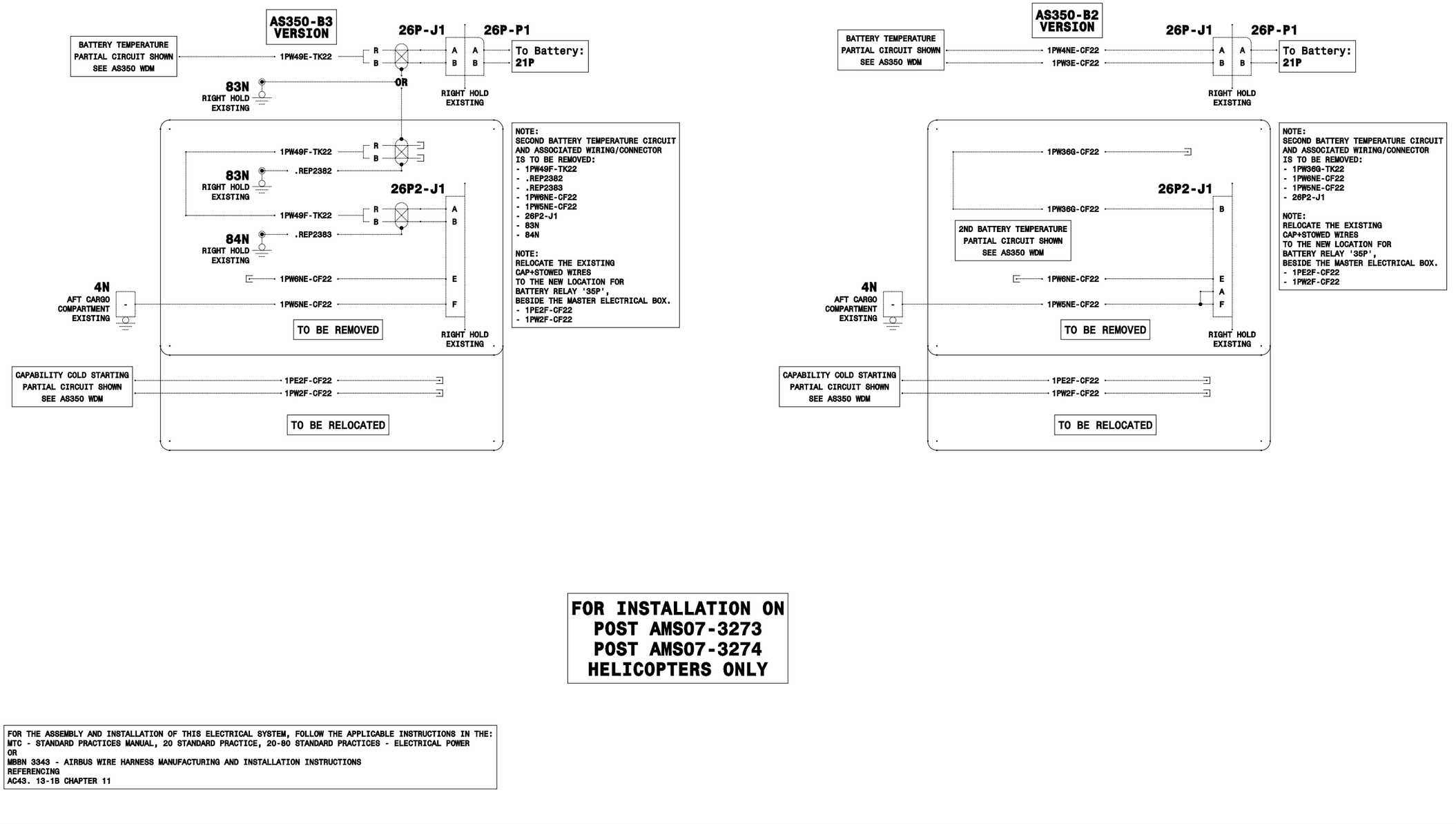


Figure 19 AS 350 Battery Relocation Page 2 of 6, Wiring Diagram



FOR INSTALLATION ON
POST AMS07-3273
POST AMS07-3274
HELICOPTERS ONLY

FOR THE ASSEMBLY AND INSTALLATION OF THIS ELECTRICAL SYSTEM, FOLLOW THE APPLICABLE INSTRUCTIONS IN THE:
MTC - STANDARD PRACTICES MANUAL, 20 STANDARD PRACTICE, 20-80 STANDARD PRACTICES - ELECTRICAL POWER
OR
MBBN 3343 - AIRBUS WIRE HARNESS MANUFACTURING AND INSTALLATION INSTRUCTIONS
REFERENCING
AC43. 13-1B CHAPTER 11

Figure 20 AS 350 Battery Relocation Page 3 of 6, Wiring Installation

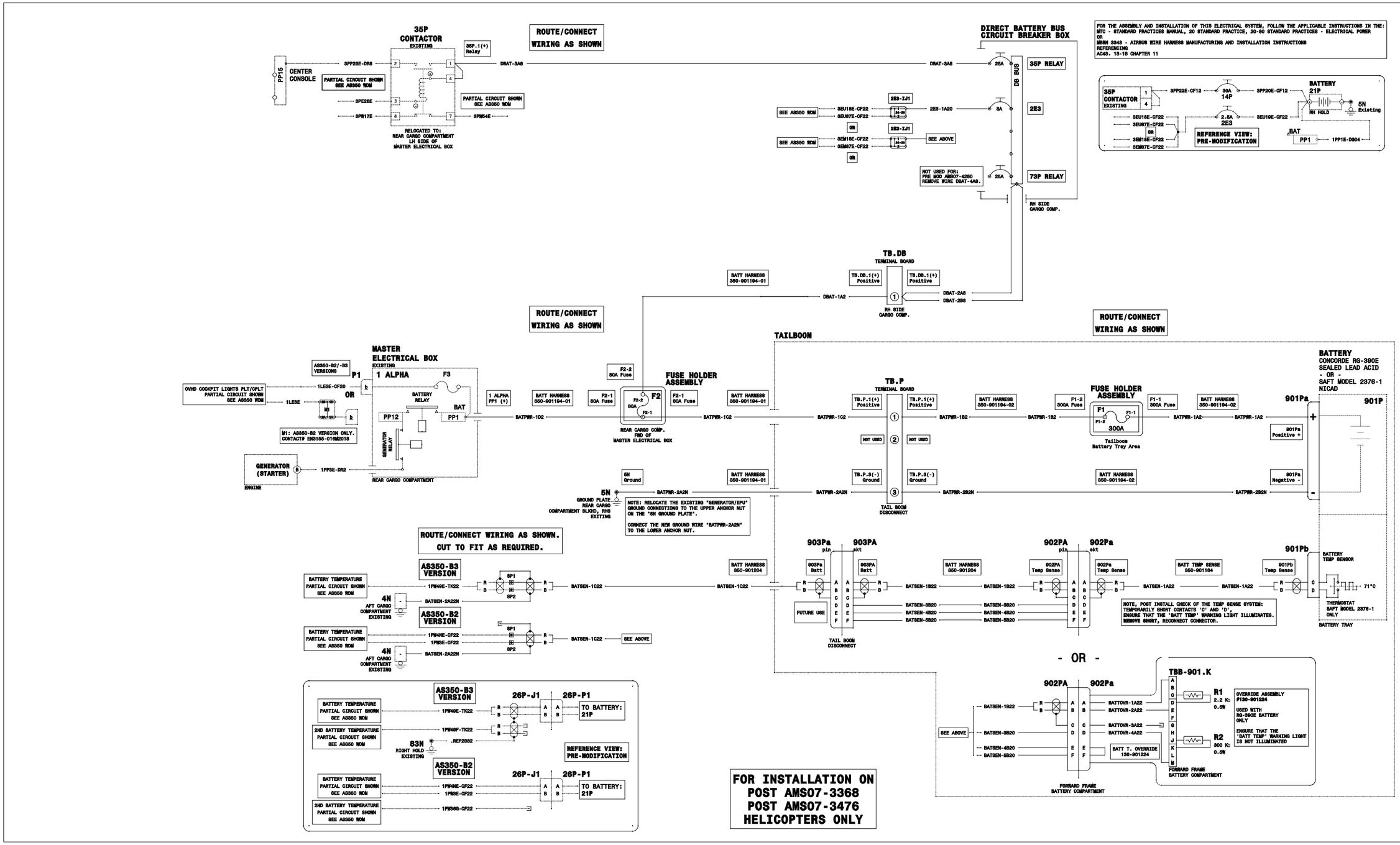


Figure 21 AS 350 Battery Relocation, Page 4 of 6, Wiring Diagram

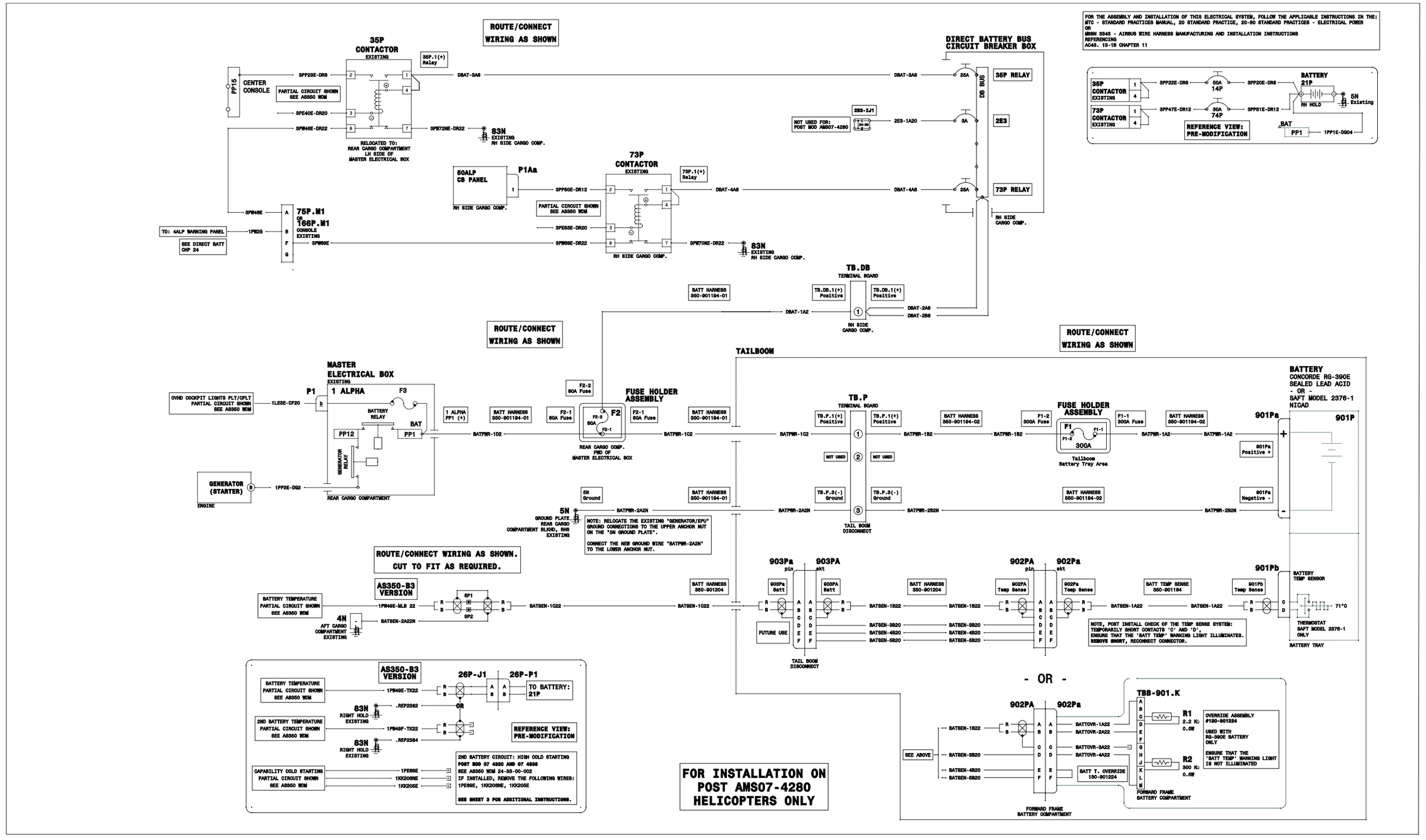


Figure 22 AS 350 Battery Relocation, Page 5 of 6, Wiring Diagram

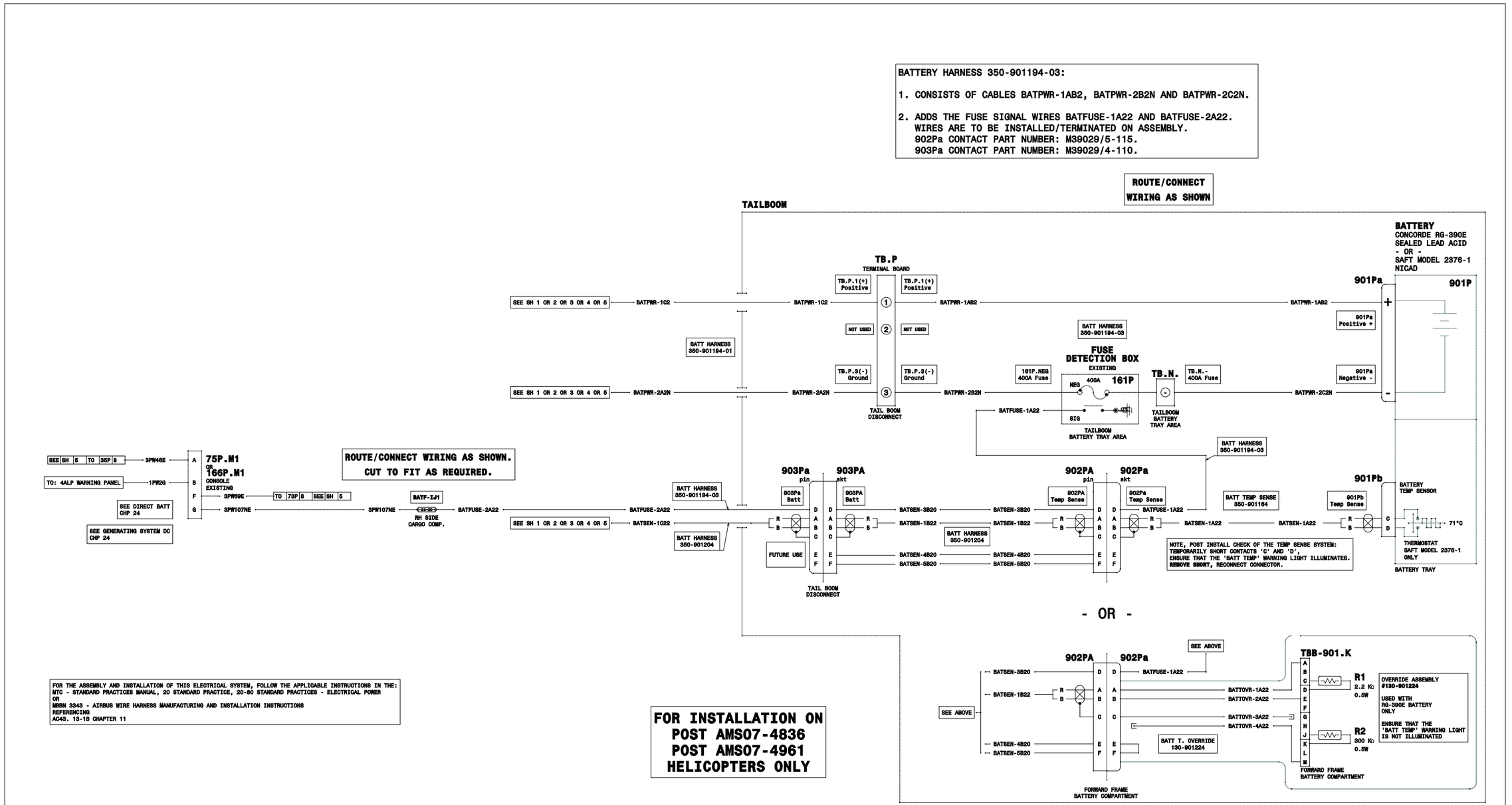
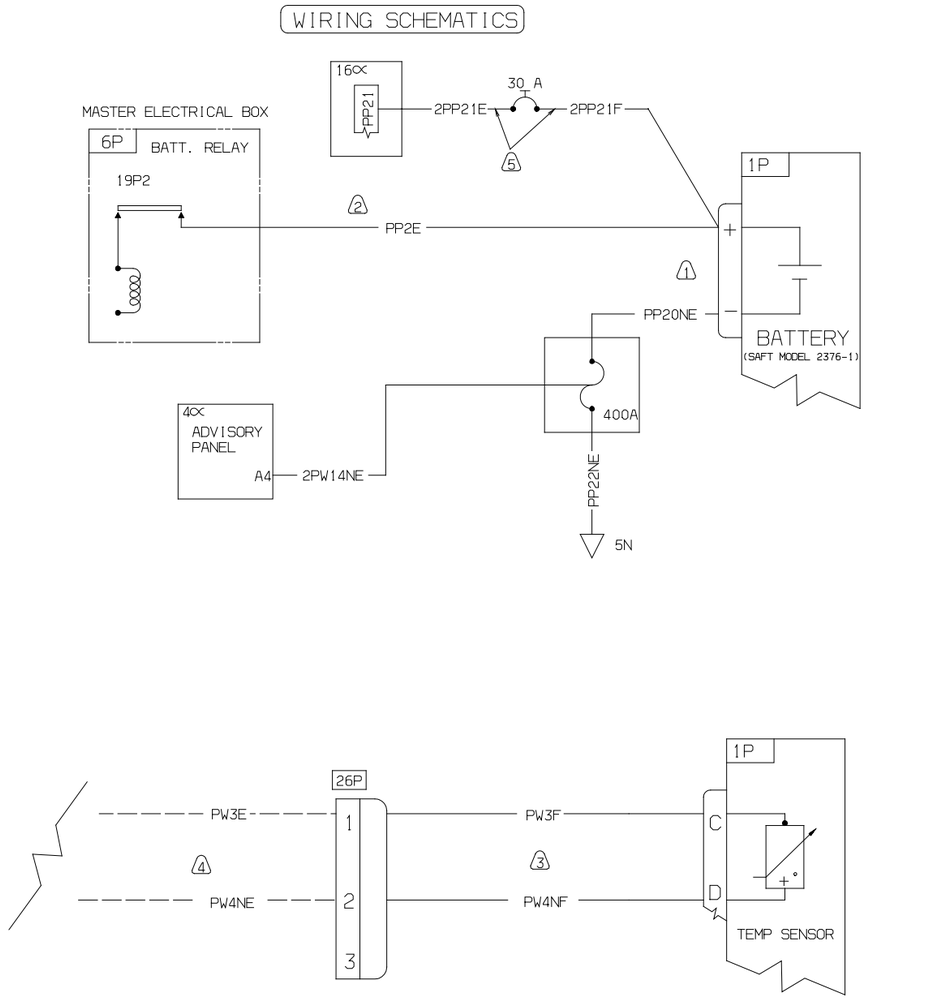


Figure 23 AS 350 Battery Relocation, Page 6 of 6, Wiring Diagram



5. WIRE SIZE 8AWG
4. ROUTE EXISTING TEMP SENSOR WIRES TOWARDS TAILBOOM
3. WIRE SIZE 20 AWG UNLESS OTHERWISE SPECIFIED
2. WIRE SIZE 1 AWG UNLESS OTHERWISE SPECIFIED
1. REFER TO CHAPTER 24.30.00. REMOVE EXISTING BATTERY HOT LIVE PPIE AND BONDING STRAP.

NOTES:

Figure 24 AS 355 Battery Relocation, Page 1 of 1, Wiring Diagram

Left blank intentionally.

7 SPECIAL TOOLING

For special tooling requirements for the Concorde Battery, refer to the Concorde Battery Corporation, Component Maintenance Manual, Document Number 5-0171, CMM 24-30-71, Rev. Q, Dec., 4, 2018 (or latest revision). For special tooling requirements for the SAFT Battery, refer to the SAFT Operating and Maintenance Manual, Document Number OMM 24-30-99, February 14, 2024 (or latest version).

8 REMOVAL AND REPLACEMENT

PRELIMINARIES

A. For AS 350 (excluding AS 350 B2/B3):

- Read General Safety Instructions - Electrical Power Supply System, refer to AS 350 MET, Chapter 24-00-00-301.
- Comply with Instructions Applicable during Maintenance, refer to MTC, Chapter 20-07-03-401.
- Disconnect the external power in accordance with AS 350 MET, Chapter 24-00-00-301 (if applicable).
- Open and secure applicable circuit breakers / fuse before any servicing action.

B. For AS 350 B2/B3:

- Read General Safety Instructions - Electrical Power Supply System, refer to AS 350 AMM, Chapter 24-00-00, 3-1.
- Comply with Instructions Applicable during Maintenance, refer to MTC, Chapter 20-07-03-401.
- Disconnect the external power in accordance with AS 350 AMM, Chapter 24-00-00, 2-1 (if applicable).
- Open and secure applicable circuit breakers / fuse before any servicing action.

C. For AS 355:

- Read General Electrical Power System, AS 355 MET, Chapter 24-00-00-301.
- Comply with Instructions Applicable during maintenance, refer to AS 355 MTC Chapter 20-07-03-401.
- Disconnect the external power in accordance with AS 355 MET, PRE MOD 07-4280 or POST MOD 074280, Chapter 24-00-00-301, (if applicable).
- Open and secure applicable circuit breakers / fuse before any servicing action.

D. Open battery compartment access door located in tail boom LH side by releasing the two camlocks.

NOTE: Follow safety precautions in the Concorde Battery Corporation, Component Maintenance Manual, Document Number 5-0171, CMM 24-30-71, Rev. Q, Dec., 4, 2018 (or latest version) and SAFT Operating and Maintenance Manual, Document Number OMM 24-30-99, February 14, 2024 (or latest version) before removing/installing battery.

8 REMOVAL AND REPLACEMENT (continued)

A. REMOVAL

General Repair Instruction Unriveting principle - refer to MTC, Chapter 20-03-01-102.

1) BATTERY (Refer to Figure 2 and 5)

a) Disconnect battery connector (11). If nickel-cadmium battery is installed, disconnect the ttemperature plug (2).

b) Disconnect FWD and AFT vent hoses (3 and 4) from the quick connection (19) on the battery base.

NOTE: If replacing fwd or aft vent hose remove tyrap (5) and disconnect hose clamp (4). Remove hose. Refer to Figure 5.

c) Looosen both FWD and AFT locking knobs (5) on the battery tray assembly (6). Refer to Figure 2..

d) Carefully slide the battery tray assembly (6) out of the tail boom.

e) Remove u-plate (7) by loosening the self-locking nut (8) and remove the battery clamp assembly (9).

f) Carefully remove battery (10) from battery tray assembly (6).

NOTE: Follow storage procedures in Concorde Battery Corporation, Component Maintenance Manual, Document Number 5-0171, CMM 24-30-71, Rev. Q, Dec. 4, 2018 and SAFT Operating and Maintenance Manual, Document Number OMM 24-30-99, February 14, 2024.

2) BATTERY CONNECTOR (Refer to Figure 2)

a) Remove bolts securing battery harness (1) to the battery connector (11). Retain bolts for reinstallation.

3) ACCESS DOOR SEAL (Refer to Figures 14, 15 and 16)

a) Remove the self-locking nuts (3, 6 places), washers (4, 6 places) and screws (6, 6 places) securing the access door (14) to the tail boom. Refer to Figure 16.

b) Remove access door (14) and place on workbench..

c) Carefully remove damaged seal (2) from around inside of access door (1). Refer to Figures 14 and 15.

4) CIRCUIT BREAKERS (PRE & POST MOD 07-3273 and 07-3274) (Refer to Figure 3)

a) Remove screws securing terminals to circuit breaker and remove both circuit breakers (3).

5) OUTER SKIN DOUBLER (Refer to Figure 16)

a) Remove Tail Gear Box. Refer to Removal TGB, hub body and control plate, AS 350/AS 355 (excluding B2/B3) MET, Chapter 65-20-00,401.

Remove TGB-Tail Gear Box. Refer to Removal-TGB-Tail Gear Box, AS 350 B2/B3 AMM, Chapter 65-21-00,4-1.

b) Remove Tail Rotor Drive Shaft. Refer to Removal Tail Rotor Drive Shaft, AS 350 (excluding B2/B3) MET, Chapter 65-10-00-401.

For AS 350 B2/B3, refer to Removal - Front Shaft Section-Tail rotor drive, AS 350 B2/B3 AMM, Chapter 65-11-00, 4-1, Removal- Rear Section-Tail rotor drive line, AS 350 B2/B3 AMM, Chapter 65-21-00,4-5.

c) Support tail boom.

d) Remove the battery following instructions in Section 8.A.1) BATTERY.

e) Remove access door to the tail boom in accordance with Section 8.A.3)a). Place access door on work bench.

f) Disconnect harnesses and secure out of working area.

g) Remove rivets securing base (7) onto battery box.

h) Remove rivets securing outer skin doubler (7) to the tail boom.

NOTE: Removing rivets will also remove inner skin doubler (8).

8 REMOVAL AND REPLACEMENT (continued)

B. REPLACEMENT

NOTE: Use torque per MTC, Chapter 20-02-05-404, unless otherwise specified.

NOTE: Deburr all sharp corners and edges. Alodine and prime all reworked surfaces.

NOTE: Finish paint all reworked surfaces as required to match original surface colour.

Replacement of rivets - MTC, Chapter 20-03-02-101.

General Sealing procedures – MTC, Chapter 20-05-01-101

General methods of applying sealing compounds – MTC, Chapter 20-05-01-102

Application of PR 1422-B2 Sealant - MTC, Chapter 20-05-01-206

General rule for bonding with adhesives - MTC, Chapter 20-06-01-101

Bonding with adhesives - MTC, Chapter 20-06-01-102

1) CIRCUIT BREAKERS (PRE & POST MOD 07-3273 and 07-3274) (Refer to Figure 3)

a) Connect terminals to both circuit breakers (3) and secure using screws.

2) OUTER SKIN DOUBLER (Refer to Figure 5, 8, 9, 10, 11, 12, 13 and 16)

a) Temporarily secure outer skin doubler (7) to tail boom and match drill any existing holes from tail boom into outer skin doubler (7). Remove doubler and deburr. Refer to Figure 16.

b) If replacing inner skin doubler (8), temporarily position inner skin doubler (8) onto tail boom skin. Locate tail boom seam and trim inner doubler to clear upper tail boom skin. Refer to NOTE 2.

c) Match drill any existing holes from tail boom into inner skin doubler (8). Remove doubler and deburr.

NOTE: Countersink holes in outer skin doubler (7) for hinges (1, 2 places) and receptacles (2, 2 places).

d) Reposition skin doubler(s) (7 and 8) onto tail boom. Secure upper and lower stiffeners (12 & 13) to tail boom using rivets (47 places, MS20426AD4-6).

e) Secure LH Frame (15) at STN 1817 to tail boom using rivets (28 places, MS20470AD4-5).

f) Secure frame at STN 2287 to tail boom using rivets (29 places, MS20470AD4-5).

g) Secure around opening using rivets (31 places, MS20426AD4-8).

h) Secure upper and lower TC stiffener (9, 2 places) to tail boom using rivets (18 places, MS20470AD4-6).

i) Secure skin doublers (7 and 8) forward of opening using rivets (19 places, MS20470AD4-5). Secure outer skin doubler (7) at tail boom seam using rivets (3 places, MS20470AD4-5).

j) Secure skin doublers (7 and 8) aft of opening using rivets (31, MS20470AD4-5)

k) Reinstall both receptacles using rivets (4 places, MS20426AD3-8).

l) Bead seal edges of doublers (7 and 8) with sealant (17). Refer to NOTE 1.

m) Reposition base (7) onto battery box and secure:

For AS 350 PRE MOD 07-3273 and 07-3274 secure using rivets (11, 8 places) and rivets (MS20470AD4-4, 24 places). Refer to Figure 8.

8 REMOVAL AND REPLACEMENT (continued)

B. REPLACEMENT (continued)

2) OUTER SKIN DOUBLER (Refer to Figure 5, 6, 7, 8, 9, 10, 11, 12, 13 and 16) (continued)

For AS 350 POST MOD 07-3273 and 07-3274 secure using rivets (11, 8 places) and rivets (MS20470AD4-4, 42 places). Refer to Figure 9

For AS 350 POST MOD 07-4280 (AS 350 B3) secure using rivets (11, 8 places) and rivets (MS20470AD4-4, 42 places). Refer to Figure 10

For AS 350 PRE MOD 07-4836 - 01 Variant secure using rivets (13, 8 places), rivets (MS20426AD4-5, 20 places) and rivets (MS20426AD4-6, 4 places). Refer to Figure 11.

For AS 350 POST MOD 07-4836 - 02 Variant secure using rivets (14, 8 places), rivets (MS20426AD4-5, 20 places) and rivets (MS20426AD4-6, 4 places). Refer to Figure 12.

For AS 355 secure using rivets (MS20426AD4-4, 24 places) and rivets (MS20470AD4-4, 24 places). Refer to Figure 13.

n) Reposition access door and secure using self-locking nuts (3, 6 places), washers (4, 6 places) and screws (6, 6 places). Refer to Figure 16.

o) Install Tail Rotor Gear Box. Refer to Installation TGB, hub body and control plate, AS 350/AS 355 (excluding B2/B3) MET, Chapter 65-20-00-401.

Install Tail Rotor Gear Box. Refer to Installation-TGB-Tail Gear Box, AS 350 B2/B3 AMM, Chapter 65-21-00,4-2.

p) Install the Tail Rotor Drive Shaft. Refer to Installation-TGB-Tail Rotor Drive Shaft, AS 350 (excluding B2/B3) MET, Chapter 65-10-00-401.

Install the Tail Rotor Drive Shaft. Refer to Installation-Front section-Tail rotor drive line, AS 350 B2/B3, AMM, Chapter 65-11-00, 4-2, Installation-Rear section-Tail rotor drive line, AS 350 B2/B3 AMM Chapter 65-11-00,4-6.

q) Reposition Battery following instructions given in 8.B.4) BATTERY.

r) Reconnect harnesses.

3) ACCESS DOOR SEAL (Refer to Figures 14, 15 and 16)

a) Trim neoprene self-adhesive (2) as necessary. Remove backing and apply to access door (1). Refer to Figure 14.

b) Trim P-Seal (2) as necessary. Apply adhesive (6) to door and secure P-Seal (2). Refer to Figure 15.

c) Reposition access door (14) and secure using self-locking nuts (3, 6 places), washers (4, 6 places) and screws (6, 6 places). Refer to Figure 16.

4) BATTERY (Refer to Figure 2)

a) Carefully place battery (10) on battery tray assembly (6) and ensure correct seating.

b) Install the battery clamp assembly (9, 2 places) and secure by sliding the u-plate (7) down until it meets the battery (10) and secure using the self-locking nut (8).

c) Slide battery tray assembly (6) into position and secure with the two locking knobs (5).

d) Connect battery temperature plug (2, used only if nickel-cadmium battery is installed) to battery.

e) Connect FWD and AFT vent hoses (3 and 4) to the quick connect (19) on the battery base.

NOTE: If fwd or aft vent hose is being replaced, ensure hose is correct length. Once cut to fit, secure to elbow (3) using hose clamp (4). Slide hose over hose barb (6) and secure using tywrap (5). Refer to Figure 5.

5) BATTERY CONNECTOR (Refer to Figure 2)

a) Secure battery harness (1) to the battery connector (11) using previously retained bolts.

8 REMOVAL AND REPLACEMENT (continued)

B. REPLACEMENT (continued)

6) Use an ohm meter, point to point check all connections to ensure correct installation.

7) Close all areas opened for service in the PRELIMINARIES paragraph of this section.

For AS 350 (excluding AS 350 B2/B3):

- Connect battery.
- Reconnect the external power unit, AS 350 MET, Chapter 24-00-00-301 (If required)
- Reference functional test - DC Power system in accordance with AS 350 MET, Chapter 24-30-00-501.

For AS 350 B2/B3:

- Connect battery
- Reconnect the external power unit, AS 350 B2/B3 AMM, Chapter 24-00-00, 2-1. (if required)
- Reference functional test - DC Power Supply System in accordance with AS 350 B2/B3, AMM, Chapter 24-30-00-5-1.

For AS 355:

- Connect battery.
- Reconnect external power unit in accordance with AS 355 MET, Chapter 24-00-00-301, PRE MOD 07-4280, or POST MOD 07-4280 (if required).
- Reference functional test - DC Generating System in accordance with AS 355 MET, Chapter 24-30-00-501.

8) For Lead Acid battery only:

- On power up, ensure that the “BATT TEMP” warning light is not illuminated.

For NiCad Battery only:

- Temporarily short contact “C” and “D” of connector “901Pb”, ensure that the “BATT TEMP” Warning light illuminates. Remove the short and reconnect the connector.

9) Close the battery compartment access door and secure door using the two camlocks.

10) Perform operational check of all systems that were serviced in accordance with the AS 350/ 355 MET/AMM procedures and the system's installation/operation manual.

9 WEIGHT AND BALANCE DATA

NOTE: Refer to SB No. AS350-53.00.43 for MOD OP-4309 to improve tail boom thermal protection on lengthy hover flight maneuvers for the AS 350 BB3.

Lead Acid Battery (Concorde) Installation

A. Removed Items

DESCRIPTION	WEIGHT		ARM		MOMENT	
	kg	lbs	m	in	kg m	lb in
Battery (existing)	-15.00	-33.07	3.85	151.57	-57.75	-5012.42
Tray, tail boom skin and hardware (existing)	-2.00	-4.41	3.85	151.57	-7.70	-668.42
Total	-17.00	-37.48	3.85	151.57	-65.45	-5680.84

B. Added Items

DESCRIPTION	WEIGHT		ARM		MOMENT	
	kg	lbs	m	in	kg m	lb in
Lead Acid Battery (Concorde)	28.12	61.99	7.15	281.50	201.06	17450.19
Tray, structural support, access door, harness and hardware	16.00	35.27	7.15	281.50	114.40	9928.51
Total	44.12	97.26	7.15	281.50	315.46	27378.70

Nickel-Cadmium Battery (SAFT) Installation

A. Removed Items

DESCRIPTION	WEIGHT		ARM		MOMENT	
	kg	lbs	m	in	kg m	lb in
Battery (existing)	-15.00	-33.07	3.85	151.57	-57.75	-5012.42
Tray, tail boom skin and hardware (existing)	-2.00	-4.41	3.85	151.57	-7.70	-668.42
Total	-17.00	-37.48	3.85	151.57	-65.45	-5680.84

B. Added Items

DESCRIPTION	WEIGHT		ARM		MOMENT	
	kg	lbs	m	in	kg m	lb in
Nickel-Cadmium Battery (SAFT)	25.00	55.12	7.15	281.50	178.75	15516.28
Tray, structural support, access door, harness and hardware	16.00	35.27	7.15	281.50	114.40	9928.51
Total	41.00	90.39	7.15	281.50	293.15	25444.79

10 **PLACARDS AND MARKINGS**

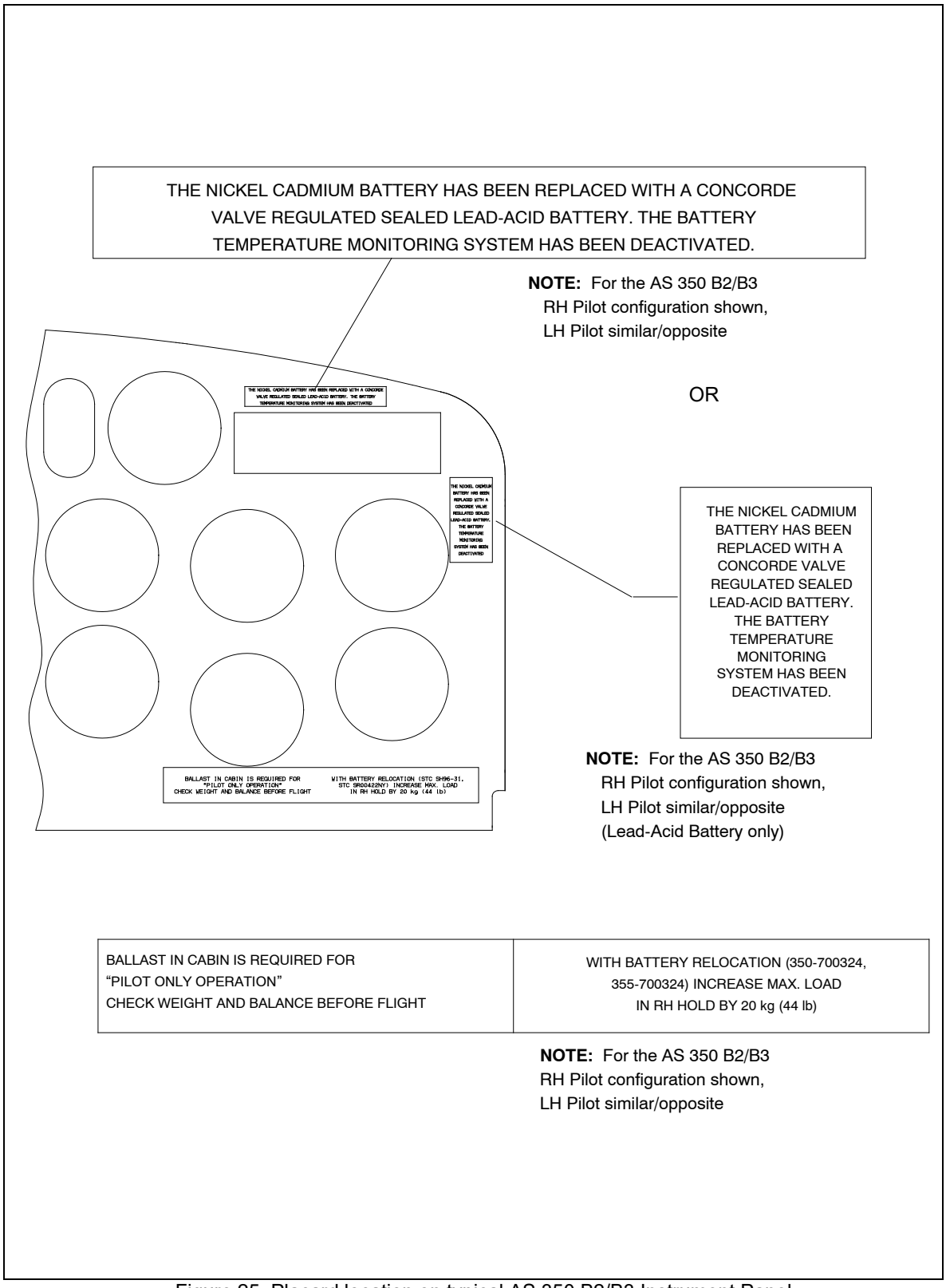
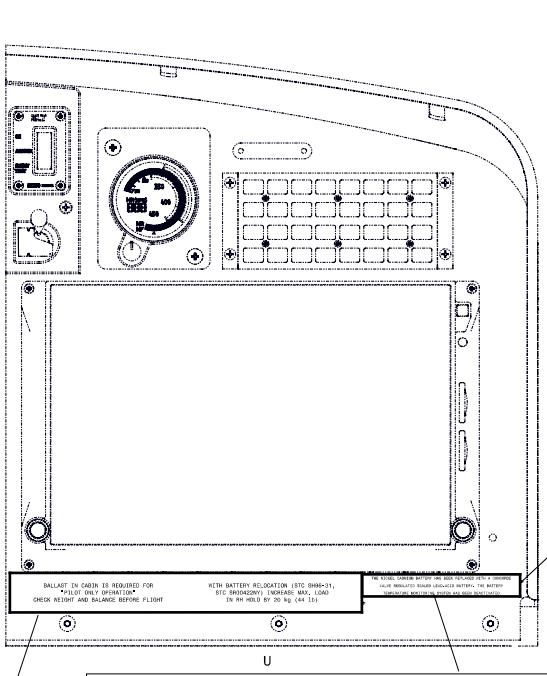


Figure 25 Placard location on typical AS 350 B2/B3 Instrument Panel

10 **PLACARDS AND MARKINGS**



THE NICKEL CADMIUM BATTERY HAS BEEN REPLACED WITH A CONCORDE VALVE REGULATED SEALED LEAD-ACID BATTERY. THE BATTERY TEMPERATURE MONITORING SYSTEM HAS BEEN DEACTIVATED.

BALLAST IN CABIN IS REQUIRED FOR "PILOT ONLY OPERATION" CHECK WEIGHT AND BALANCE BEFORE FLIGHT	WITH BATTERY RELOCATION (350-700324, 355-700324) INCREASE MAX. LOAD IN RH HOLD BY 20 kg (44 lb)

Figure 26 Placard location on AS 350 B3 Step 3 Instrument Panel

10 **PLACARDS AND MARKINGS** (continued)

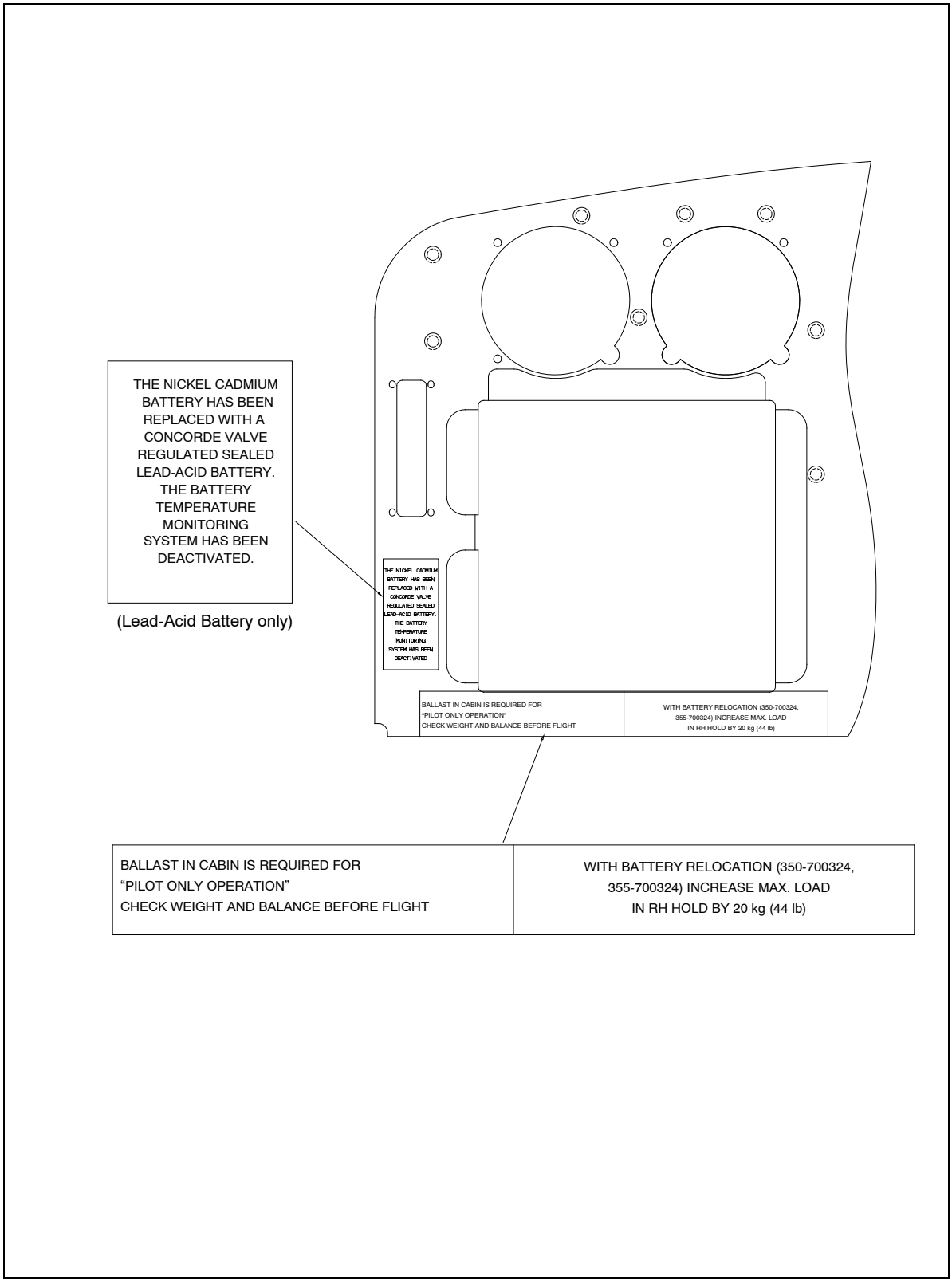
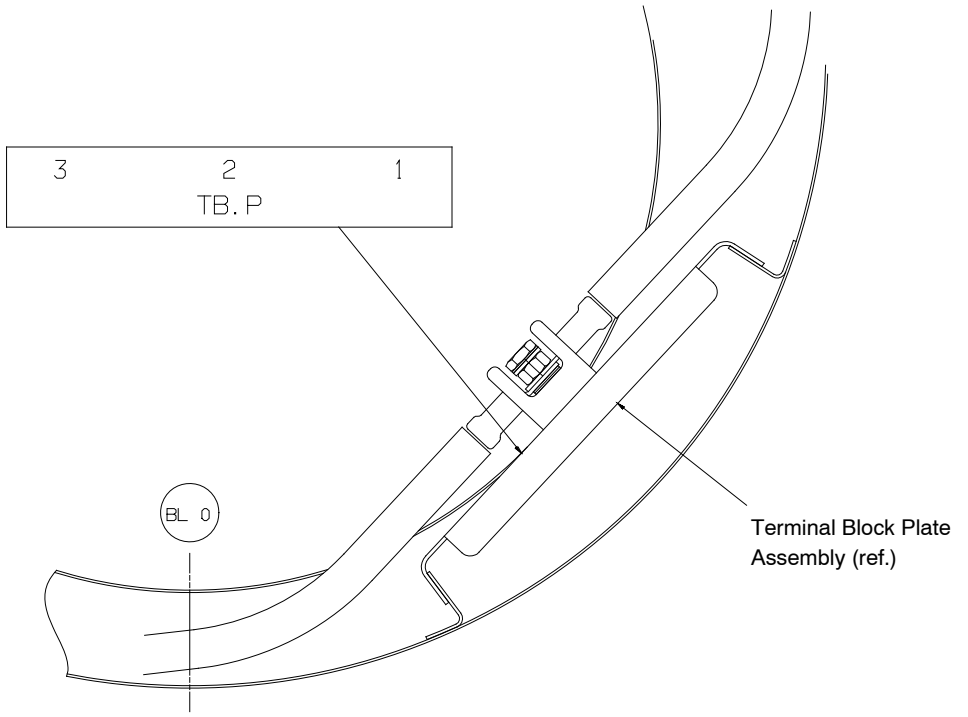


Figure 27 Placard location on typical AS 355 Instrument Panel

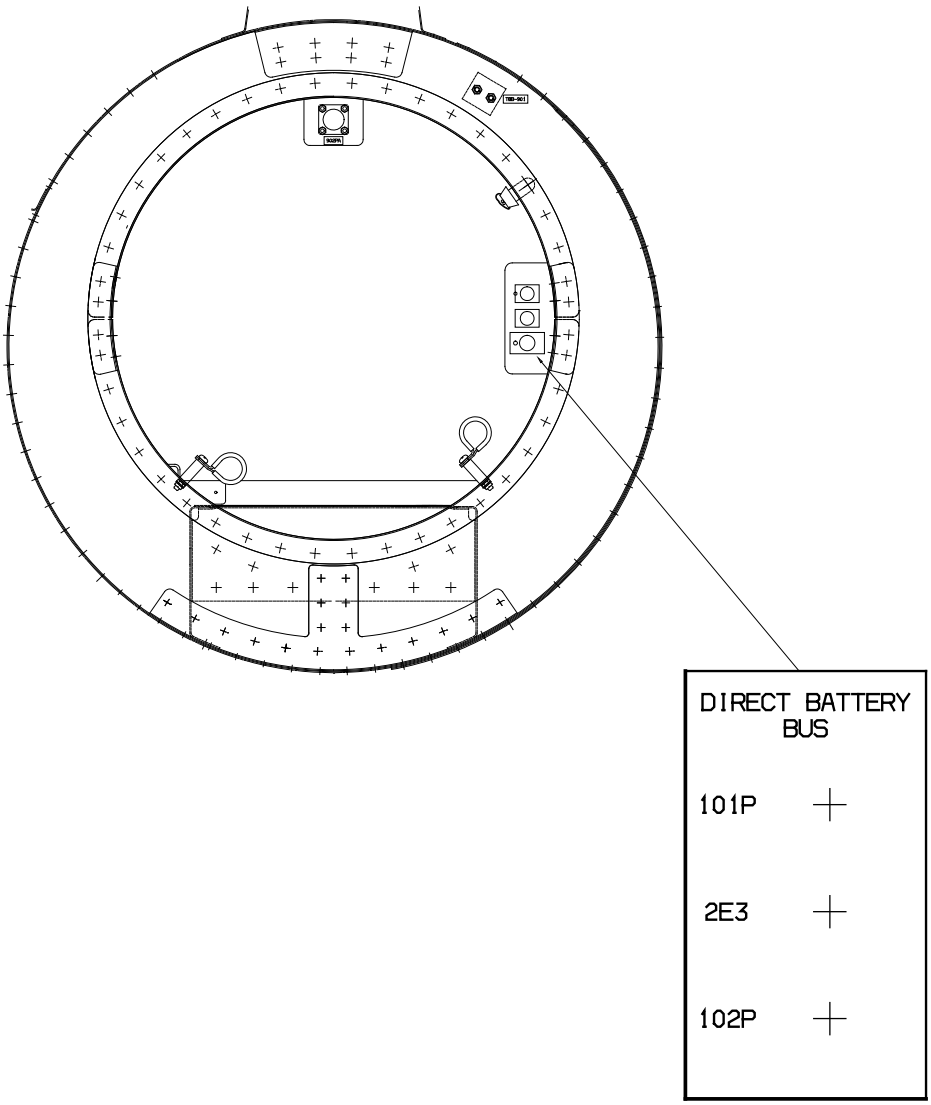
10 **PLACARDS AND MARKINGS** (continued)



NOTE: For the AS 350

Figure 28 Marking location on typical AS 350 terminal block plate assembly for the terminal block plate (TB.P) at STN A156

10 **PLACARDS AND MARKINGS** (continued)



NOTE: For the AS 350

Figure 29 Marking location on the AS 350 circuit bracket at STN A1817
POST MOD 07-3273 and 07-3274

10 **PLACARDS AND MARKINGS** (continued)

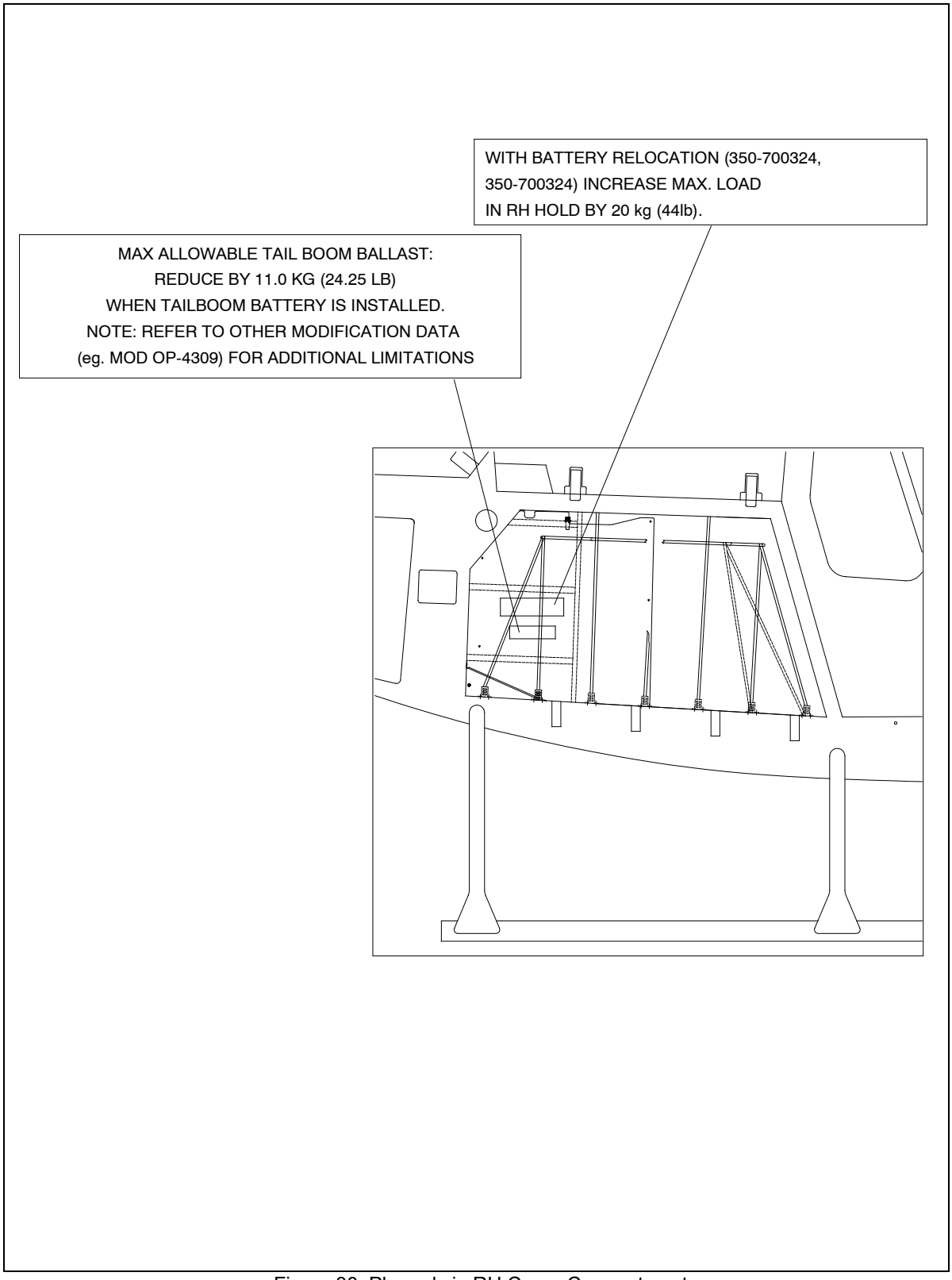


Figure 30 Placards in RH Cargo Compartment

10 **PLACARDS AND MARKINGS** (continued)

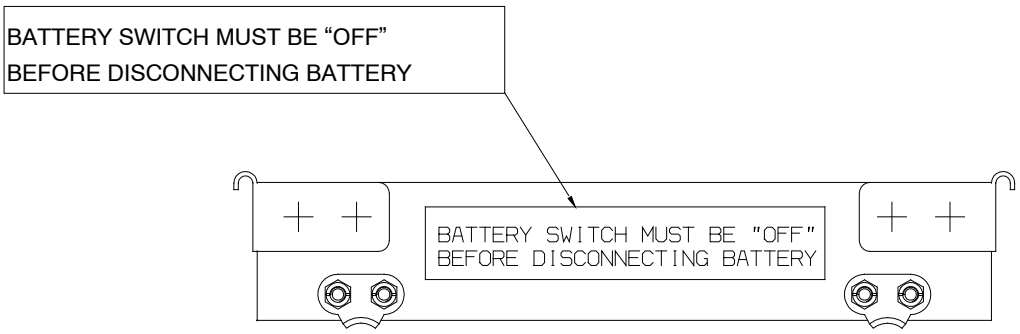


Figure 31 Placard on front of the Battery Tray Assembly